

METRIC/INCH-POUND

**KSC-DE-512-SM
REVISION L**

**FACILITY SYSTEMS, GROUND SUPPORT SYSTEMS,
AND GROUND SUPPORT EQUIPMENT
GENERAL DESIGN REQUIREMENTS**

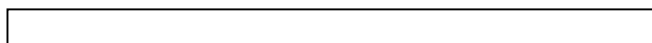
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ENGINEERING AND TECHNOLOGY DIRECTORATE

National Aeronautics and
Space Administration

John F. Kennedy Space Center

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GENERAL DESIGN REQUIREMENTS**

Approved by:

Patrick A. Simpkins
Director, Engineering and Technology Directorate

This Revision Supersedes All Previous Editions of This Document

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JOHN F. KENNEDY SPACE CENTER, NASA

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RECORD OF REVISIONS/CHANGES			
REV LTR	CHANGE NO.	DESCRIPTION	DATE
		Basic issue.	January 1983
A		General revision.	March 1986
	A-1	Added requirements for SCAPE suit operations and EMI compatibility.	February 20, 1987
B		General revision.	June 1988
	B-1	Added requirements for marking of test weights.	July 26, 1991
C		General revision.	March 10, 1993
D		Miscellaneous changes.	August 5, 1994
	D-1	Added requirements for design of facility premises wiring.	December 15, 1994
E		Added requirements for instrumentation calibration and quick release pins.	June 1, 1995
F		Revised miscellaneous references.	August 5, 1996
G		Updated miscellaneous references and text.	December 18, 1998
H		Updated miscellaneous references and text.	September 14, 2000
	H-1	Revised 3.3.3.2.13.	May 30, 2002
J		Updated miscellaneous references and text. Incorporated change H-1.	October 15, 2002
K		Updated miscellaneous references and text. All pages reformatted.	September 30, 2004
L		Complete rewrite and align with NASA-STD-5005 .	February 29, 2012

FOREWORD

KSC-DE-512-SM establishes overall requirements and best design practices to be used at the John F. Kennedy Space Center (KSC) for the development of ground systems (GS) in support of operations at launch, landing, and retrieval sites. These requirements apply to the design and development of hardware and software for ground support equipment (GSE), ground support systems (GSS), and facility ground support systems (F-GSS) used to support the KSC mission for transportation, receiving, handling, assembly, test, checkout, servicing, and launch of space vehicles and payloads and selected flight hardware items for retrieval.

During the 1950s and early 1960s, the Missile Firing Laboratory (later renamed to the Launch Operations Directorate) was the launch operations arm of Redstone Arsenal and the Army Ballistic Missile Agency. The Missile Firing Laboratory used Army specifications and standards for its design and development of ground systems. KSC's effort to develop standards began with GP-863, General Criteria for Design of New Equipment and Facilities, which was released in July 1970 and updated three years later. GP-863 focused on operability, reliability, maintainability, useful life, environmental, transportability, human performance, safety, logistics, documentation, and quality assurance. KSC-DE-512-SM, Facility Systems, Ground Support Systems, and Ground Support Equipment, General Design Requirements, replaced GP-863 in 1983. The early revisions of KSC-DE-512-SM contained requirements, along with guidance for accomplishing detailed designs. Later revisions became more formal and the "shall" statement became the phrase to identify each requirement.

This standards manual supplements [NASA-STD-5005](#) by including KSC-site-specific and local-environment requirements. KSC-DE-512-SM is a single, complete document for design and development of KSC ground systems for use at launch, landing, and retrieval sites.

These requirements and practices are optional for equipment used at manufacturing, development, and test sites.

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FACILITY SYSTEMS, GROUND SUPPORT SYSTEMS, AND GROUND SUPPORT EQUIPMENT GENERAL DESIGN REQUIREMENTS

1. SCOPE

1.1 Purpose

KSC-DE-512-SM establishes requirements and guidance for design and fabrication of ground systems (GS) that includes: ground support equipment (GSE), ground support systems (GSS), and facility ground support systems (F-GSS) to provide uniform methods and processes for design and development of robust, safe, reliable, maintainable, supportable, and cost-effective GS in support of space flight and institutional programs and projects. This standard is intended to supplement the minimum requirements of [NASA-STD-5005](#) by applying more stringent, restrictive, or demanding requirements applicable to the specific KSC environment. This standard also provides requirements for GSS and F-GSS because these systems are not covered in [NASA-STD-5005](#).

1.2 Applicability

The applicability of KSC-DE-512-SM relative to other NASA and industry design standards is shown graphically in [Figure 1](#).

- a. This standards manual applies to all new GS for programs and projects assigned to KSC.
- b. KSC-DE-512-SM establishes minimum design requirements for ground systems, as defined herein, for NASA programs and projects assigned to KSC. This standards manual is intended to establish uniform engineering best practices and methods in the design, documentation, procurement, fabrication, assembly, test, and installation of ground systems to support KSC operations.
- c. This standards manual is required for use by KSC design entities and their support contractors and may be cited in contracts, projects, and other documents as necessary to provide a technical requirement.
- d. Rationale and guidance are provided in italic text after the requirement where more definition is needed.
- e. The requirements of this standard are optional for hardware used only at manufacturing, development, or test sites but are required for hardware used at launch, landing, and retrieval sites.
- f. A program may invoke additional requirements that differ from the requirements stated herein. These requirements will be evaluated and approved for use by the KSC Engineering and Technology Directorate Technical Authority. See [KSC-PLN-5400](#) and paragraph [1.3](#).

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- h. KSC Directorates are responsible to invoke this standards manual for design and development of ground systems. KSC Directorates have responsibilities described as follows:
 - (1) Engineering and Safety and Mission Assurance (S&MA) have the responsibility to determine categories or types of GS (e.g., critical vs. noncritical) and any additional requirements resulting from these categories or types.
 - (2) Center Operations, Protective Services Branch, Authority Having Jurisdiction, has the responsibility and authority to define hazardous areas and approve designs that provide equipment to these areas.
- i. This standards manual applies to the following ground systems (see paragraph 3.2):
 - (1) ground support equipment,
 - (2) ground support systems,
 - (3) facility ground support systems,
 - (4) special test equipment, and
 - (5) modifications to commercial off-the-shelf (i.e., modified off-the-shelf [MOTS]).
- j. [NASA-STD-5005](#) applies to GSE provided by entities other than KSC.
- k. This standards manual does not apply to the following:
 - (1) tools (standard shop),
 - (2) facilities and utilities (in-house and architect & engineering [A&E] firms), or
 - (3) commercial off-the-shelf (COTS) equipment.

KSC-DE-512-SM does not cover the design and fabrication of tools, facilities and utilities, and COTS equipment. However, other KSC standards and industry standards may apply to these designs.

- l. The Technical Authority determines the classification of equipment listed in paragraphs [1.2.i](#), [1.2.j](#), and [1.2.k](#).

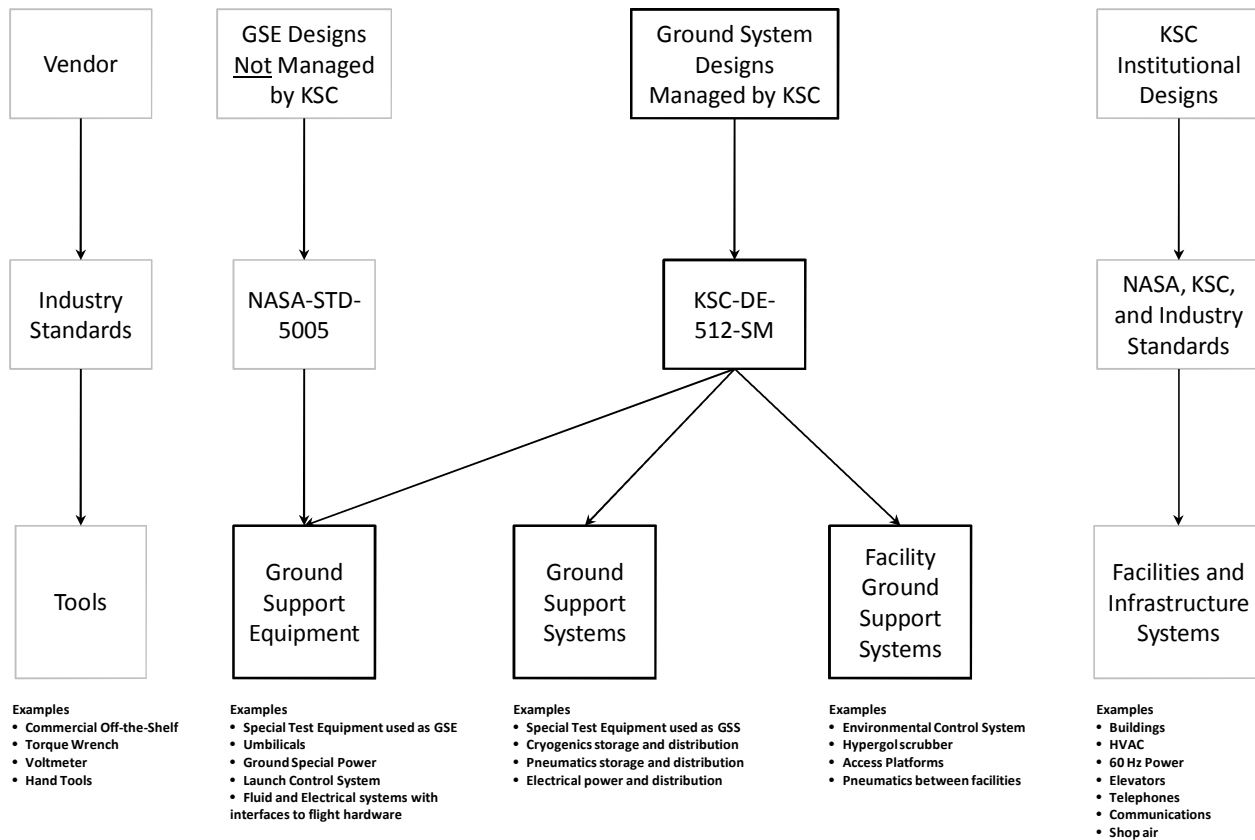


Figure 1. KSC-DE-512-SM Applicability

1.3 Tailoring and Waivers

Individual provisions of this standards manual should be and are intended to be tailored (i.e., modified or deleted) to meet specific program and project needs. All tailoring shall be evaluated for use by the KSC Engineering Directorate Technical Authority (see [KSC-PLN-5400](#)). Waivers to institutional requirements shall follow the process in [KDP-KSC-P-1865](#).

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2. APPLICABLE DOCUMENTS

2.1 General

The latest issuances of documents listed in this section contain provisions that constitute requirements of this standard as cited in the text.

The applicable documents are accessible via the NASA Standards and Technical Assistance Resource Tool at <http://standards.nasa.gov> or may be obtained directly from the standards developing organizations or other document distributors.

2.2 Government Documents

Department of Defense

AFSPCMAN 91-710	Range Safety User Requirements Manual, Volume 3, Launch Vehicles, Payloads, and Ground Support Systems Requirements
MIL-DTL-16878	Wire, Electrical, Insulated, General Specification for
MIL-DTL-22992	Connectors, Plugs and Receptacles, Electrical, Waterproof, Quick Disconnect, Heavy Duty Type, General Specification for
MIL-DTL-24308	Connectors, Electric, Rectangular, Nonenvironmental, Miniature, Polarized Shell, Rack and Panel, General Specification for
MIL-DTL-38999	Connectors, Electrical, Circular, Miniature, High Density, Quick Disconnect (Bayonet, Threaded and Breech Coupling), Environment Resistant, Removable Crimp and Hermetic Solder Contacts, General Specification for
MIL-HDBK-17-1	Composite Materials Handbook, Volume 1, Polymer Matrix Composites Guidelines for Characterization of Structural Materials
MIL-HDBK-17-2	Composite Materials Handbook, Volume 2, Polymer Matrix Composites Materials Properties
MIL-HDBK-17-3	Composite Materials Handbook, Volume 3, Polymer Matrix Composites Materials Usage, Design, and Analysis

MIL-HDBK-17-4	Composite Materials Handbook, Volume 4, Metal Matrix Composites
MIL-HDBK-17-5	Composite Materials Handbook, Volume 5, Ceramic Matrix Composites
MIL-HDBK-149	Rubber
MIL-HDBK-263	Electrostatic Discharge Control Handbook for Protection of Electrical and Electronic Parts, Assemblies and Equipment (Excluding Electrically Initiated Explosive Devices) (Metric)
MIL-HDBK-700	Plastics
MIL-HDBK-6870	Inspection Program Requirements, Nondestructive for Aircraft and Missile Materials and Parts
MIL-PRF-39012	Connectors, Coaxial, Radio Frequency, General Specification for
MIL-STD-202	Test Method Standard, Electronic and Electrical Component Parts
MIL-STD-461	Requirements for the Control of Electromagnetic Interference Characteristics of Subsystems and Equipment
MIL-STD-464	Electromagnetic Environmental Effects Requirements for Systems
MIL-STD-810	Environmental Engineering Considerations and Laboratory Tests
MIL-STD-889	Dissimilar Metals
MIL-STD-1686	Electrostatic Discharge Control Program for Protection of Electrical and Electronic Parts, Assemblies and Equipment (Excluding Electrically Initiated Explosive Devices)
<u>Federal (FED)</u>	
29 CFR 1910	Occupational Safety and Health Standards
29 CFR 1926	Safety and Health Regulations for Construction

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49 CFR 171 through 180	Hazardous Materials Regulations (Department of Transportation)
FED-STD-595	Colors Used in Government Procurement
HF-STD-001	Human Factors Design Standard (Federal Aviation Administration)
DOT/FAA/AR-MMPDS	Metallic Materials Properties Development and Standardization (MMPDS)
<u>NASA</u>	
NASA-HDBK-1001	Terrestrial Environment (Climatic) Criteria Handbook for Use in Aerospace Vehicle Development
NASA-SPEC-5004	Welding of Aerospace Ground Support Equipment and Related Nonconventional Facilities
NASA-STD-4003	Electrical Bonding for NASA Launch Vehicles, Spacecraft, Payloads, and Flight Equipment
NASA-STD-5005	Standard for the Design and Fabrication of Ground Support Equipment
NASA-STD-5008	Protective Coating of Carbon Steel, Stainless Steel, and Aluminum on Launch Structures, Facilities, and Ground Support Equipment
NASA-STD-5009	Nondestructive Evaluation Requirements for Fracture-Critical Metallic Components
NASA-STD-6001	Flammability, Offgassing, and Compatibility Requirements and Test Procedures
NASA-STD-6002	Applying Data Matrix Identification Symbols on Aerospace Parts
NASA-STD-8719.9	Standard for Lifting Devices and Equipment
NASA-STD-8719.12	Safety Standard for Explosives, Propellants, and Pyrotechnics
NASA-STD-8719.17	NASA Requirements for Ground-Based Pressure Vessels and Pressurized Systems (PV/S)

NASA-STD-8739.1	Workmanship Standard for Polymeric Application on Electronic Assemblies
NASA-STD-8739.2	Workmanship Standard for Surface Mount Technology
NASA-STD-8739.3	Soldered Electrical Connections
NASA-STD-8739.4	Crimping, Interconnecting Cables, Harnesses, and Wiring
NASA-STD-8739.5	Fiber Optic Terminations, Cable Assemblies, and Installation
NPR 2810.1	Security of Information Technology
NPR 6000.1	Requirements for Packaging, Handling, and Transportation for Aeronautical and Space Systems, Equipment, and Associated Components
NPR 7150.2	NASA Software Engineering Requirements
NPR 8705.2	Human-Rating Requirements for Space Systems
NPR 8715.3	NASA General Safety Program Requirements
<u>George C. Marshall Space Flight Center (MSFC), NASA</u>	
MSFC-SPEC-445	Adhesive Bonding, Process and Inspection, Requirements for
MSFC-STD-486	Threaded Fasteners, Torque Limits, Standard for
MSFC-STD-3029	Guidelines for the Selection of Metallic Materials for Stress Corrosion Cracking Resistance in Sodium Chloride Environments
<u>John F. Kennedy Space Center (KSC), NASA</u>	
120E3100003	Electrical Cable Fabrication Requirements
79K07491	Installation of Purge Hardware
79K80000 Series	Fluid Component Specification Drawings
KDP-KSC-P-1865	Institutional Requirement Deviation/Waiver Process

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KSC-PLN-5400	Technical Authority and Technical Excellence Implementation Plan
KNPR 8700.2	KSC System Safety and Reliability Analysis Methodology Procedural Requirements
KNPR 8715.3	KSC Safety Practices Procedural Requirements
KNPR 8720.2	KSC Reliability and Maintainability Procedural Requirements
KNPR 8730.1	KSC Metrology and Calibration Procedural Requirements
KNPR 8730.2	Quality Assurance Procedural Requirement
KSC-C-123	Surface Cleanliness of Ground Support Equipment Fluid Systems, Specification for
KSC-DD-818-TR	Summary of Measurements of KSC Launch-Induced Environmental Effects (STS-1 through STS-11)
KSC-DF-107	Technical Documentation Style Guide
KSC-E-165	Electrical Ground Support Equipment Fabrication, Specification For
KSC-E-166	Electrical Ground Support Equipment, Installation and Assembly, Specification for
KSC-GP-425	Fluid Fitting Engineering Standards
KSC-GP-435	Engineering Drawing Practices, Vol. 1: Aerospace and Ground Support Equipment
KSC-GP-864	Electrical Ground Support Equipment Cable Handbook
KSC-GP-986	Design Criteria for Reusable Space Vehicle Umbilical Systems
KSC-NE-9187	Sensors, Transducers and Signal Conditioning Systems Selection Guidelines
KSC-SPEC-E-0002	Modular Electrical Enclosures, Racks, Consoles, and Accessories, Specification for

KSC-SPEC-P-0012	Refractory Concrete, Specification for
KSC-SPEC-P-0027	Tubing, Steel, Superaustenitic, Corrosion Resistant, UNS N08367 and UNS S31245, Seamed, Bright Annealed, Passivated, Specification for
KSC-SPEC-Z-0007	Tubing, Steel, Corrosion Resistant, Types 304 and 316, Seamless, Annealed, Specification for
KSC-SPEC-Z-0008	Fabrication and Installation of Flared Tube Assemblies and Installation of Fittings and Fitting Assemblies, Specification for
KSC-SPEC-Z-0009	Lubrication, Thread, Corrosion-Resistant Steel and Aluminum Alloy Tube Fittings, Specification for
KSC-STD-132	Potting and Molding Electrical Cable Assembly Terminations, Standard for
KSC-STD-141	Load Test Identification and Data Marking, Standard for
KSC-STD-164	Environmental Test Methods for Ground Support Equipment, Standard for
KSC-STD-E-0001	Design of Electrical Control and Monitoring Systems, Equipment (GSE), and Panels, Standard for
KSC-STD-E-0002	Hazardproofing of Electrically Energized Equipment, Standard for
KSC-STD-E-0004	Pneumatic and Hydraulic Mechanical Components, Electrical Design, Standard for
KSC-STD-E-0011	Electrical Power Receptacles and Plugs, Standard for
KSC-STD-E-0012	Facility Grounding and Lightning Protection, Standard for
KSC-STD-E-0015	Standard for Marking of Ground Support Equipment
KSC-STD-G-0003	Launch Support and Facility Components, Qualification of, Standard for

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KSC-STD-P-0006	Quick Release Pins and Pin Tethers, Standard for
KSC-STD-SF-0004	Ground Piping Systems Color Coding and Identification, Safety Standard for
KSC-STD-Z-0005	Pneumatic Ground-Support Equipment, Design of, Standard for
KSC-STD-Z-0006	Hypergolic Propellants Ground Support Equipment, Design of, Standard for
KSC-STD-Z-0008	Ground Life Support Systems and Equipment, Design of, Standard for
KSC-STD-Z-0009	Cryogenic Ground Support Equipment, Design of, Standard for
KSC-STD-Z-0010	Environmental Control Systems, Ground Coolant Systems, Coolant Servicing Systems, and Ground Support Equipment, Design of, Standard for
MMA-1985-79	Standard Test Method for Evaluating Triboelectric Charge Generation and Decay

2.3 Non-Government Documents

Aerospace Industries Association (AIA)/National Aerospace Standards (NAS)

AIA/NAS 410	NAS Certification and Qualification of Nondestructive Test Personnel
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Aluminum Association (AA)

ADM	Aluminum Design Manual
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American Institute of Steel Construction (AISC)

AISC 325	Steel Construction Manual
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American National Standards Institute (ANSI)

ANSI C18.2M Part 1	American National Standard for Portable Rechargeable Cells and Batteries – General and Specifications
ANSI C18.3M Part 1	American National Standard for Portable Lithium Primary Cells and Batteries – General and Specifications

American Society for Testing and Materials (ASTM)

ASTM A123	Standard Specification for Zinc (Hot-Dip Galvanized) Coatings on Iron and Steel Products
ASTM A153	Standard Specification for Zinc Coating (Hot-Dip) on Iron and Steel Hardware
ASTM A312	Standard Specification for Seamless, Welded, and Heavily Cold Worked Austenitic Stainless Steel Pipes
ASTM A325	Standard Specification for Structural Bolts, Steel, Heat Treated, 120/105 ksi Minimum Tensile Strength
ASTM A380	Standard Practice for Cleaning, Descaling, and Passivation of Stainless Steel Parts, Equipment, and Systems
ASTM A490	Standard Specification for Structural Bolts, Alloy Steel, Heat Treated, 150 ksi Minimum Tensile Strength
ASTM A653	Standard Specification for Steel Sheet, Zinc-Coated (Galvanized) or Zinc-Iron Alloy-Coated (Galvannealed) by the Hot-Dip Process
ASTM A780	Standard Practice for Repair of Damaged and Uncoated Areas of Hot-Dip Galvanized Coatings
ASTM A967	Standard Specification for Chemical Passivation Treatments for Stainless Steel Parts
ASTM B241	Standard Specification for Aluminum and Aluminum-Alloy Seamless Pipe and Seamless Extruded Tube
ASTM D7194	Standard Specification for Aerospace Parts Machined from Polychlorotrifluoroethylene (PCTFE)
ASTM E1417	Standard Practice for Liquid Penetrant Testing
ASTM E1444	Standard Practice for Magnetic Particle Testing

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ASTM E1548	Standard Practice for Preparation of Aerospace Contamination Control Plans
ASTM E1742	Standard Practice for Radiographic Examination
ASTM E2217	Standard Practice for Design and Construction of Aerospace Cleanrooms and Contamination Controlled Areas
ASTM E2375	Standard Practice for Ultrasonic Testing of Wrought Products
ASTM MNL 36	Safe Use of Oxygen and Oxygen Systems: Handbook for Design, Operation, and Maintenance

American Society of Civil Engineers (ASCE)

ASCE-7	Minimum Design Loads for Buildings and Other Structures
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American Society of Mechanical Engineers (ASME)

ASME BPVC-VIII	ASME Boiler & Pressure Vessel Code, Section VIII, Division 1: Rules for Construction of Pressure Vessels, Division 2: Alternative Rules, Division 3: Alternative Rules
ASME BPVC-X	ASME Boiler & Pressure Vessel Code, Section X, Fiber-Reinforced Plastic Pressure Vessels
ASME Y14.100	Engineering Drawing Practices
ASME B30.1	Jacks, Industrial Rollers, Air Casters, and Hydraulic Gantries
ASME B31.3	Process Piping
ASME B31.8	Gas Transmission and Distribution Piping Systems
ASME B31.9	Building Services Piping

American Welding Society (AWS)

AWS C3.2M/C3.2	Standard Method for Evaluating the Strength of Brazed Joints
AWS C3.4M/C3.4	Specification for Torch Brazing

AWS C3.5M/C3.5	Specification for Induction Brazing
AWS C3.6M/C3.6	Specification for Furnace Brazing
AWS C3.7M/C3.7	Specification for Aluminum Brazing
<u>Compressed Gas Association (CGA)</u>	
CGA C-7	Guide to the Preparation of Precautionary Labeling and Marking of Compressed Gas Containers
<u>ESD Association</u>	
ESD S20.20	For the Development of an Electrostatic Discharge Control Program for – Protection of Electrical and Electronic Parts, Assemblies and Equipment (Excluding Electrically Initiated Explosive Devices)
<u>International Electrotechnical Commission</u>	
IEC 60807	Rectangular Connectors for Frequencies Below 3 MHz
<u>IPC – Association Connecting Electronics Industries</u>	
IPC-2221	Generic Standard on Printed Board Design
IPC-2222	Sectional Design Standard for Rigid Organic Printed Boards
IPC-2223	Sectional Design Standard for Flexible Printed Boards
IPC-2252	Design Guide for RF/Microwave Circuit Boards
IPC-6011	Generic Performance Specification for Printed Boards
IPC-6012	Qualification and Performance Specification for Rigid Printed Boards
IPC-6013	Qualification and Performance Specification for Flexible Printed Boards
IPC-6018	Microwave End Product Board Inspection and Test
IPC J-STD-001	Requirements for Soldered Electrical and Electronic Assemblies, Performance Class 3

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IPC J-STD-001ES Space Applications Electronic Hardware Addendum
to IPC J-STD-001E Requirements for Soldered
Electrical and Electronic Assemblies

National Fire Protection Association (NFPA)

NFPA 59A Standard for the Production, Storage, and Handling
of Liquefied Natural Gas (LNG)

NFPA 70 National Electrical Code

NFPA 70E Standard for Electrical Safety in the Workplace

NFPA 496 Standard for Purged and Pressurized Enclosures for
Electrical Equipment

Sheet Metal and Air Conditioning Contractors' National Association (SMACNA)

SMACNA 1958 HVAC Systems Duct Design

Society of Automotive Engineers (SAE)

SAE AMS-DTL-23053/5 Insulation Sleeving, Electrical, Heat Shrinkable,
Polyolefin, Flexible, Crosslinked

SAE AMS 2175 Castings, Classification and Inspection of

SAE AMS 2403 Plating, Nickel, General Purpose

SAE AMS 2404 Plating, Electroless Nickel

SAE AMS 2423 Plating, Nickel Hard Deposit

SAE AMS 2460 Chromium Plating

SAE AMS 2488 Anodic Treatment – Titanium and Titanium Alloys
Solution pH 13 or Higher

SAE AMS 2647 Fluorescent Penetrant Inspection Aircraft and
Engine Component Maintenance

SAE AMS 2759 Heat Treatment of Steel Parts, General
Requirements

SAE AMS 2759/9 Hydrogen Embrittlement Relief (Baking) of Steel
Parts

SAE AMS 2770 Heat Treatment of Wrought Aluminum Alloy Parts

SAE AMS 2771	Heat Treatment of Aluminum Alloy Castings
SAE AMS 2772	Heat Treatment of Aluminum Alloy Raw Materials
SAE AMS 2774	Heat Treatment Wrought Nickel Alloy and Cobalt Alloy Parts
SAE AMS-H-6875	Heat Treatment of Steel Raw Materials
SAE AMS-H-81200	Heat Treatment of Titanium and Titanium Alloys
SAE AMS-QQ-P-416	Plating, Cadmium (Electrodeposited)
SAE AMS-STD-2154	Inspection, Ultrasonic, Wrought Metals, Process for
SAE ARP 1247	Aerospace Ground Support Equipment – General Requirements
SAE ARP 4402	Eddy Current Inspection of Open Fastener Holes in Aluminum Aircraft Structure
SAE AS 4787	Eddy Current Inspection of Circular Holes in Nonferrous Metallic Aircraft Engine Hardware
SAE AS 5942	Marking of Electrical Insulating Materials
SAE AS 8090	Mobility, Towed Aerospace Ground Equipment, General Requirements for
SAE AS 22759	Wire, Electrical, Fluoropolymer-Insulated, Copper or Copper Alloy
SAE AS 50151	Connectors, Electrical, Circular Threaded, AN Type, General Specification for
SAE AS 50861	Wire, Electric, Polyvinyl Chloride Insulated, Copper or Copper Alloy

2.4 Order of Precedence

This standard establishes requirements and guidance for design and fabrication of KSC ground systems (GSE, GSS, and F-GSS), but does not supersede or waive established Agency requirements found in other documentation. Conflicts between this standard and applicable documents cited herein shall be resolved by the responsible Technical Authorities.

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3. ACRONYMS, ABBREVIATIONS, AND DEFINITIONS

3.1 Acronyms and Abbreviations

°C	degrees Celsius
°F	degrees Fahrenheit
A-50	aerozine 50
AA	Aluminum Association
AC	alternating current
ADM	Aluminum Design Manual
AFSPCMAN	Air Force Space Command Manual
AIA	Aerospace Industries Association
AIAA	American Institute of Aeronautics and Astronautics
AISC	American Institute of Steel Construction
AMS	Aerospace Material Specification
ANSI	American National Standards Institute
ARP	Aerospace Recommended Practice
AS	Aerospace Standard
ASCE	American Society of Civil Engineers
ASME	American Society of Mechanical Engineers
ASTM	American Society for Testing and Materials
AWS	American Welding Society
BPVC	Boiler and Pressure Vessel Code
CFR	Code of Federal Regulations
CGA	Compressed Gas Association
CIL	Critical Items List
COPV	Composite Overwrapped Pressure Vessel
COTS	commercial off-the-shelf
DE	design
DOT	Department of Transportation
DTL	detail
ECS	Environmental Control System
ECLSS	Environmental Control and Life Support System
ECA	Electronic Components, Assemblies & Materials Association
ECS	Environmental Control System
EEE	electrical, electronic, and electromechanical
e.g.	for example
EIA	Electronic Industries Association
EMC	electromagnetic compatibility
EMI	electromagnetic interference
EPS	Electrical Power System
ESD	electrostatic discharge
ETFE	ethylene tetrafluoroethylene
FAA	Federal Aviation Administration
FED	Federal
F-GSS	facility ground support system

FMEA	Failure Mode and Effects Analysis
FOD	foreign-object debris
GO ₂	gaseous oxygen
GP	general publication (KSC)
GS	ground system
GSE	ground support equipment
GSS	ground support system
HDBK	handbook
HVAC	heating, ventilation, and air conditioning
Hz	hertz
ICD	interface control document
i.e.	that is
IEC	International Electrotechnical Commission
IEEE	Institute of Electrical and Electronic Engineers
in	inch
IPC	Association Connecting Electronics Industries
ISO	International Standardization Organization for Standardization
IT	information technology
JSC	Lyndon B. Johnson Space Center
kPa	kilopascal
KSC	John F. Kennedy Space Center
ksi	one thousand pounds per square inch
KTI	Kennedy Technical Instruction
LNG	liquefied natural gas
M&P	materials and processes
MAPTIS	Materials and Processes Technical Information System
MHz	megahertz
MIL	military
mm	millimeter
MMA	Malfunction/Materials Analysis
MMH	monomethylhydrazine
MMPDS	Metallic Materials Properties Development and Standardization
MNL	manual
MPa	megapascal
MSFC	George C. Marshall Space Flight Center
MUA	Material Usage Agreement
N ₄ H ₄	hydrazine
N ₂ O ₄	nitrogen tetroxide
NAS	National Aerospace Standards
NASA	National Aeronautics and Space Administration
NDE	nondestructive evaluation
NEMA	National Electrical Manufacturers Association
NFPA	National Fire Protection Association
NPD	NASA Policy Directive
NPR	NASA Procedural Requirements

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NSS	NASA Safety Standard
OSHA	Occupational Safety and Health Administration
PCTFE	polychlorotrifluoroethylene
PFA	plastic film, foam, and adhesive tape
pH	potential of hydrogen
PHE	propellant handler's ensemble
PPE	personal protective equipment
PRF	performance specification
psi	pound per square inch
psia	pound per square inch absolute
PTFE	polytetrafluoroethylene
PV/S	pressure vessels and pressurized systems
QD	quick disconnect
RF	radio frequency
RH	relative humidity
RP	reference publication
S&MA	Safety and Mission Assurance
SAA	Systems Assurance Analysis
SAE	Society of Automotive Engineers
SCC	stress corrosion cracking
SMACNA	Sheet Metal and Air Conditioning Contractors' National Association
SPEC	specification
SSP	Space Shuttle Program
STD	standard
TM	technical memorandum
TP	technical procedure
UDMH	unsymmetrical dimethylhydrazine
UNS	Unified Numbering System
UTS	ultimate tensile strength
vs.	versus

3.2 Definitions

analysis: use of calculations, numerical simulations, tools, techniques, and physics/engineering-based modeling to determine that the requirement is satisfied.

Rationale: Engineering analysis proceeds by separating the engineering design into the components and disciplines, analyzing, or estimating each component of the operation or failure mechanism separately, and recombining the components in accordance with physics and engineering principles. The methods selected must be supported by appropriate technical rationale and be documented in detail.

catastrophic: a level beyond critical, resulting in loss of life, loss of flight vehicle, or loss of a major ground asset.

commercial off-the-shelf (COTS): equipment, including hardware and associated software/procedures, that is commercially available from the industrial inventory at the time of purchase.

Rationale: Commercial items or components should be used when they satisfy the ground systems function and will not degrade the safety or reliability of the ground or flight system. Requirements should be specified in terms of functionality or performance rather than design. To qualify as COTS, equipment must not be modified (see modified off-the-shelf [MOTS]).

conventional structures: structures composed predominately of standard structural shapes and connections typical of commercial or residential construction, such as office buildings, warehouses, machine shops, and other facilities whose structures are characterized by well-established design precedents and loading conditions.

corrosive environment: a marine (sea coast) or launch-induced environment that causes degradation of materials due to oxidation or chemical reaction.

Rationale: In the corrosive marine (sea coast) environment at KSC, the most common sources of corrosion are moisture and sodium chloride. The launch-induced environment also introduces hydrochloric acid, which exacerbates the corrosive effects of the marine environment.

critical ground system: a system whose loss of function or improper performance could result in serious injury, damage to flight hardware, loss of mission, or major damage to a significant ground asset.

critical: loss of function or improper performance could result in serious injury, damage to flight hardware, loss of mission, or major damage to a significant ground asset.

criticality: a program-defined measure of the consequences of a failure mode.

Rationale: Criticality of a ground system (GS) is determined by a Safety and Mission Assurance study analysis of the function and application of the equipment. The classifications assigned to the GS will guide the design team in determining which specifications and standards to apply, which materials to select, and how to document the GS.

demonstration: a verification method that determines the properties on an end item by observation of its operation. This method is generally employed where qualitative operational performance is to be verified. A demonstration must be witnessed and documented.

design life: the operational life of equipment (to include storage life, installed life in a nonoperating mode, and operational service life), after which the equipment will be replaced or recertified. It is the responsibility of the program/project to determine recertification requirements, which may include refurbishment, analysis, or test.

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facility: land, buildings, structures, and other real property improvements, including facility systems (utility systems and collateral equipment). Facility systems include heating, ventilation, and air conditioning (HVAC); 60-hertz (Hz) power; potable water; elevators; lighting; shop air; etc. Facility systems may support or have interfaces with ground systems. The term *facility* does not include ground support equipment, ground support systems, facility ground support systems, tools, or special test equipment.

facility ground support system (F-GSS): fixed infrastructure and equipment (not including utility systems and collateral equipment) that provides functional or physical support to GSS or GSE. F-GSS are specialized systems that are designed, built, and tested to more stringent requirements than conventional facilities and their integral facility systems.

Rationale: F-GSS includes fixed environmental control systems (ECS), breathing air systems, long-run piping systems, high-pressure gases, liquid hydrogen (LH₂) and liquid oxygen (LO₂) storage spheres, etc. The requirements for F-GSS and GSS are the same, and the distinction between the two is not always clear. The important distinction is between F-GSS and conventional facility systems. Conventional facility systems are not covered by this standard.

flight hardware: hardware intended for launch into space, including boosters, engines, payloads, and manned or unmanned components.

fracture-critical: classification of hardware where a crack could lead to a failure that results in serious injury, damage to flight hardware, loss of mission, or major damage to a significant ground asset.

ground support equipment (GSE): nonflight equipment, systems, or devices specifically designed and developed for a direct physical or functional interface with flight hardware.

Rationale: Equipment used during the manufacturing of flight hardware is not considered to be GSE. Each program defines when manufacturing ends and processing of the flight hardware begins. If manufacturing equipment is to be used after flight hardware processing begins, it must be designed to meet GSE requirements. GSE does not include tools that are designed for general use and not specifically for use on flight hardware.

ground support system (GSS): equipment or infrastructure (portable or fixed) that provides functional or physical support to GSE. It does not directly interface with flight hardware, although it may supply commodities, power, or data that eventually reaches the flight hardware after being conditioned or controlled by GSE.

Rationale: Design standards for GSS may be similar to or, at the discretion of the program/project, identical to the design standards for GSE. Protective features designed into the GSE prevent failures from propagating to flight hardware.

ground systems (GS): ground support equipment, ground support systems, and facility ground support systems.

inspection: measurement or examination of one or more characteristics of a product or service and comparison with specified requirements to determine conformity.

limited-life item: equipment or component that degrades due to operating time, cycling, or material aging and that has a shorter lifetime than the system's design life. Limited-life items require periodic replacement or refurbishment, which must be defined in design and maintenance documents.

modified off-the-shelf (MOTS): commercially available equipment, including hardware and associated software and procedures, modified in accordance with this document for a specific application in GS.

Rationale: Modification of COTS voids the design intent of the original equipment and places responsibility for performance, functionality, and reliability on the designer.

nonconventional structures: structures that are experimental, specific to a space program, or if any of the following apply:

1. Structural members and connections are predominately atypical to commercial or residential construction.
2. The structure supports or provides direct access to flight hardware (e.g., test stands, launch complexes, access platforms in operational or research facilities, towers, and similar special-purpose facilities).
3. Structures are characterized by unusual or inadequately defined loading conditions (launch acoustics, vibration, rocket exhaust, etc.), a lack of established design precedent, or frequent modifications to support changes in the operational requirements.

Rationale: The transition between conventional and nonconventional is the interface between the structure specifically designed for processing and vehicle access and the generic building structure whose purpose is the physical support of the overall facility. The transition region between conventional and nonconventional structures should be designed to meet the criteria for both types of structure.

safe working load: the maximum assigned load the device or equipment can operationally handle and maintain. This value is marked on the device indicating maximum working capacity. This is also the load referred to as "rated load" or "working load limit." If the device has never been downrated or uprated, this also is the "manufacturer's rated load."

safety factor: a constant that has been defined for yield and ultimate design criteria and that is the ratio of the yield or ultimate design loads to the limit load (the maximum allowable design load). If the safety factor is defined in terms of stress, it is the ratio of the ultimate or yield stress to the maximum design stress. In fatigue design, it is the ratio of the calculated fatigue life to the allowable design life. This standards manual specifies the minimum safety factor for GS for

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specific structural applications (e.g., pressure vessels, threaded fasteners, and aluminum structures).

Rationale: This definition is consistent between ground and flight hardware. This definition is inherently load-based. It reduces to the traditional stress-based definition in the simplest case.

special test equipment (STE): equipment designed for limited or one-time use in a variety of applications. STE is classified as GSE or GSS and designed to the requirements of this standards manual.

Rationale: Although its use is limited, STE has the potential to cause damage to flight hardware or injury to personnel. STE includes equipment traditionally known as shop aids.

testing: an activity that determines an item's ability to meet specified requirements by subjecting the item to a set of physical, chemical, environmental, or operating actions and conditions. It includes measurements taken with certified and calibrated tools in accordance with generally accepted scientific or engineering principles.

tools: equipment designed for general use in a variety of applications. Tools are calibrated, when necessary, in accordance with industry standards.

Rationale: Tools are not designed to specifically interface with flight hardware, nor are they designed to perform a function specific to flight hardware. Their design and general use in industry includes a variety of applications that may be required on flight hardware or GSE. Tools are intended for use by trained technicians and facilitate manual operations, such as torquing fasteners, cutting wire, checking electrical continuity, and verifying surface clearances. Examples of tools include torque wrenches, crow's feet, voltmeters, go/no-go gages, screwdrivers, wire cutters, and pliers.

traceability: the data, reports, and records that document the history of a product or component from the point of origin to final use.

Rationale: Documentation may include the origin of materials and parts and certification of personnel and processes during fabrication, assembly, procurement, installation, activation, verification, and validation.

validation: a two-part process to confirm first that the design requirements comply with the stakeholders' expectations and second that the final product complies with operational requirements.

Rationale: Validation answers the question, "Does it do what the stakeholders/users/customers want it to do?"

verification: proof of compliance with design solution specifications and descriptive documents. Verification may be determined by a combination of test, analysis, demonstration, and inspection.

Rationale: Verification answers the question, "Is it built according to the design?"

4. GENERAL REQUIREMENTS

4.1 General

In order to meet customer requirements, individual system and equipment design projects may need criteria that are more stringent than those specified herein. In such cases, these criteria should be determined by the responsible design organization in consultation with its customers (e.g., users and operators).

Each program/project has the responsibility to define its own policy for the acceptance of commercial-off-the shelf (COTS) equipment in ground systems (GS).

When a program/project approves the use of COTS equipment in GS, the following design requirements apply:

- a. COTS equipment shall be evaluated for acceptability from a materials and processes (M&P) standpoint (see 6).
- b. Qualification tests and inspections shall be indicated in the engineering documentation.
- c. Vendor documentation shall be provided as evidence that the requirements of this standards manual have been met.
- d. Modifications to COTS shall be performed in accordance with this standards manual.

COTS equipment should be used to the maximum extent possible when (1) it satisfies the intended function, (2) it will not degrade the safety or reliability of the flight or ground system, and (3) it provides a cost savings that exceeds possible cost increases due to unique maintenance or logistics requirements, modifications, or an increase in the complexity of the interfacing equipment. Vendor or contractor documentation and supporting test data should be incorporated into system control documents.

4.2 Characteristics

4.2.1 Performance Characteristics

4.2.1.1 GS Designed to Meet Flight Hardware Requirements

The GS design shall support the program/project-specific operational requirements of flight hardware.

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In addition to operational requirements, GS should be designed for ease of production, manufacturing, construction, and inspection. GS should be designed to minimize the complexity and frequency of maintenance. Close manufacturing tolerances should be avoided unless required by design and performance.

4.2.1.2 Ground System Degradation and Contamination

GS shall not degrade or contaminate flight systems, other GS, subsystems, or experiments while it is being used, checked out, serviced, or otherwise handled.

4.2.1.3 GS Design for Access

GS design shall include access provisions for handling, servicing, calibrating, maintaining, and replacing components and limited-life items.

GS design should provide for ease of operation, maintenance, servicing, cleaning, and inspection of hardware and software. GS fault detection and isolation should be considered based on criticality and cost of failures.

4.2.1.4 Interfaces

- a. GS shall meet the requirements of all interfaces with new or existing hardware or software as documented in interface control documents (ICDs) or any other documentation that controls interface requirements.

Interfaces should be verified by test and/or analysis. As a design goal, the number of connections at interfaces should be minimized.

- b. Fluid, mechanical, or electrical connections in close proximity shall be designed to prevent cross-connections.

Unique design configurations and clear identification marking should both be used to minimize the probability of incorrectly mating connections. Design configurations include threads, flanges, sizes, orientation (male/female, left-hand, right-hand), pins, and keys. Identification marking (see 5.6.1) should indicate function, commodities, pressure, reference designator, etc.

- c. GS shall be compatible with all facility interfaces.

An assessment may be required to determine whether it is more cost-effective to modify the facility interface or design the GS to meet the existing facility interface. Some GS interfaces directly to the facility; some GS interfaces to GSS or to facility ground support systems (F-GSS).

4.2.2 Physical Characteristics

4.2.2.1 Design Life Duration

- a. GS shall be designed for the operational life specified by program or mission requirements and identified in design drawings and maintenance documents.

Engineering documentation should specify maintenance requirements to meet design life.

- b. Existing GS that was verified to meet the GS requirements of a previous NASA program shall be considered acceptable for use without further verification to the requirements of this standards manual.

GSE requirements for previous NASA programs are defined in [SW-E-0002](#), [NASA-STD-5005](#), [SSP 50004](#), or a previous version of this standards manual.

4.2.2.2 Limited-Life Items

Items with limited life shall be identified on design drawings and annotated with the specific limitation to the life of the item.

Use of items with a projected lifetime that is less than the design life of the GS for which the items are intended should be avoided whenever possible. Elapsed time or cycle indicators should be employed to accumulate operational time or cycles for limited-life items. The age of items that are installed in a nonoperating mode should also be tracked.

4.2.2.3 Colors

- a. The following colors shall be used for the type of GS indicated.

Color	Color Chip Number	Ground System
Gray	26440 or 26251	Electrical/electronic, hydro/pneumatic consoles, racks, and cabinets
Gray	16187 or 16473	Structural steel/aluminum
Red	11105 or 21105	Remove-before-flight items, safety equipment, and protective equipment
White	17875 or 27875	White room or clean room equipment
Black	37038	Panel lettering
Yellow or White	13538, 17875, or 27875	Handling and transportation equipment
Blue	25102	Control racks and consoles

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Color	Color Chip Number	Ground System
Yellow Brown Band (optional)	13655 (yellow) 10080 (brown)	Connections and interfaces for hypergolic fuel servicing (see KSC-STD-SF-0004 for color coding of piping systems)
Green Brown Band (optional)	14110 (green) 10080 (brown)	Connections and interfaces for hypergolic oxidizer servicing (see KSC-STD-SF-0004 for color coding of piping systems)

- b. Colors shall be in accordance with [FED-STD-595](#).

These colors apply to GS designed and painted for NASA/KSC. If the equipment is COTS, these colors should be used as guidance to make equipment consistent with other GS.

4.2.3 Reliability

GS should be designed to minimize the probability of system failure and reduce the severity of the failure effect of the system, including failures caused by operator errors and human-system interaction. Procedures and instructions to perform and document analyses (such as Failure Mode and Effects Analysis [FMEA]/Critical Items List [CIL] or sneak circuit analysis), will be in accordance with [KNPR 8720.2](#).

4.2.3.1 Redundancy

- a. Redundant systems, subsystems, or components shall be physically oriented or separated such that the failure of one will not prevent the other from performing its intended function.
- b. Redundant systems, subsystems, or components shall be designed such that common-cause failures (e.g., contamination) do not eliminate redundancy.
- c. The design of redundant systems shall provide methods for verifying each redundant element without compromising the reliability of the redundant system.

4.2.3.2 Failure Tolerance

- a. GS, except primary structure and pressure vessels/piping/tubing in rupture mode, shall be designed to sustain a failure without causing loss of life or damage to support equipment, facilities, or flight hardware (fail-safe).

Failure tolerance is not possible for primary structure and pressure vessels/tubing in rupture mode. Safety of these systems is achieved by application of design requirements contained in this standards manual.

More stringent fault tolerance requirements (e.g., two-fault tolerant for loss of flight crew) may be applicable depending on program requirements. GS should be designed such that no single failure/inadvertent operator action results in injury or in damage to or loss of ground equipment,

flight equipment, or facilities. Failure modes/inadvertent operator actions are controlled using a systematic application of approved standards and design margins.

GS may be designed to terminate operations autonomously after the first failure or inadvertent operator action and in time to preclude any scenario that results in loss of life. This approach is consistent with the historical use of the term “fail-safe” in GS design.

- b. GS shall be designed to sustain a failure and still perform its basic function (fail-operational) when necessary to meet safety or operational requirements.

Purges, ground special power, and launch release systems are examples of systems that interface with other systems whose loss would propagate failures. Requirements for availability of system functions during launch processing may necessitate fail-operational design.

- c. GS failure modes/inadvertent operator actions with the potential for loss of flight crew shall be addressed in accordance with [NPR 8705.2](#).

4.2.3.3 Failure Propagation

GS shall be designed such that failures will not be propagated to the flight systems.

The design of GS should consider how flight hardware/software failures could propagate through the GS and affect other flight systems (vent systems, etc.).

4.2.3.4 System Safety and Reliability Analysis

- a. A criticality assessment shall be performed early in the design phase to determine the need for performing a reliability and/or safety analysis.

*The criticality of a **system** failure mode will be assigned on the basis of worst-case credible failure effect, assuming the loss of all redundancy (where applicable). This will include possible catastrophic or critical effects of system failure, including the effects of loss of hardware functions.*

- b. System inputs (including dependencies such as electrical power and pneumatic purges) and outputs (including all control and monitoring functions) that could fail with potentially catastrophic or critical consequences shall be identified as reliability critical (FMEA is required).

System inputs and outputs can be obtained from a system requirements document if available.

- c. Systems with hazards to personnel or equipment during normal or credible use scenarios shall be identified as critical (Hazard Analysis is required).

In addition, information obtained from criticality assessments is often used in determining how a system will be designed, documented, fabricated, assembled, installed, and tested.

Table 1. Criticality Categories

Criticality	Criticality Definition
1	Single failure that could result in loss of life or vehicle.
1R	Redundant hardware item, which if all failed, could cause loss of life or vehicle.
1S	Single failure in a safety or hazard monitoring system that could cause the system to fail to detect, combat, or operate when needed during the existence of a hazardous condition and could result in loss of life or vehicle.
2	Single failure that could result in loss of mission, damage to a vehicle system, or major damage to a significant ground asset.
2R	Redundant hardware item, which if all failed, could cause a loss of mission, damage to a vehicle system, or major damage to a significant ground asset.
3	All other failures.

Guidelines in Determining Criticality

- *In determining the criticality of a subsystem, emergency systems or contingency and emergency operations (e.g., fire suppression, crew escape, and abort) will not be considered as a level of redundancy.*
 - *The criticality assignment for an item whose failure affects the loading or pressure on primary structure, thermal protection, or pressure vessels will be based on the worst-case potential effect of exceeding the maximum load that the structure can withstand without deformation, rupture, or collapse.*
 - *In determining the classification of criticality categories, it will be assumed that personnel actions will be performed to activate standby redundant items, as long as requirements for detectability and time to effect are met. Manual standby redundancy must be activated by a trained operator in accordance with a preapproved written procedure.*
- d. An FMEA shall be performed in accordance with program-defined methodology on those systems and subsystems identified by the criticality analysis as having the potential to cause a criticality 1, 1S, 1R, or 2 condition. Based on the results of the FMEA, components having failure modes that could cause an undesirable event (other than criticality 3) shall be identified, categorized, and approved in accordance with the applicable agency, program, or KSC requirements. In the absence of a program requirement, the methodology of [KNPR 8700.2](#) shall be used.

All components other than criticality 3 are included in a CIL report, where they are reviewed for criteria such as qualification, failure history, tests, and verifications that will justify the design.

- e. The criticality categories shown in [Table 1](#) shall be used when the program does not identify categories.
- f. Static components, such as pressure vessels, pressure lines, cables, connectors, and wiring, shall be considered only if failure can be attributed to an active component failure.

Analysis ground rules and exceptions to program methodology or related to the scope of analysis should be clearly stated with justification.

- g. A Hazard Analysis shall be performed on those systems and subsystems determined by the criticality assessment to be a potential threat to flight systems, ground systems, or safety.
- h. Based on the results of the Hazard Analysis, those hazards that could not be eliminated by design or mitigated to an acceptable level shall require a Hazard Report.

The Hazard Report summarizes all controls and verifications that would be used to mitigate the hazard. As with the FMEA, Hazard Analyses are performed and Hazard Reports are written in accordance with the applicable program requirements. In the absence of a program requirement, the methodology of [KNRP 8700.2](#) applies.

- i. FMEA and/or Hazard Analysis including the results of the analysis shall be combined into a Systems Assurance Analysis (SAA) to summarize the risks resulting from each system/subsystem analysis.
- j. CIL and Hazard Reports shall be approved and maintained as part of the design documentation.

4.2.4 Environmental Conditions

GS shall be designed to perform in the natural and induced environments to which it will be subjected during its life cycle.

4.2.4.1 Natural Environment

GS used or stored in an uncontrolled exterior environment shall be designed to function after exposure to the natural environment at its respective geographical location as specified in [NASA-HDBK-1001](#).

Specifications in [NASA-HDBK-1001](#) may be tailored to reflect program-defined risk and exposure times, including operation within the launch commit criteria of the vehicle. Purging of enclosures to protect GS components from the natural environment should be considered essential, in addition to purging of enclosures in hazardous areas as required in [5.4.17](#).

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4.2.4.2 Launch-Induced Environment

- a. GS required to function during or after exposure to the launch-induced environment shall be designed to withstand the environment defined in program-specific documents.
- b. Testing of GS for launch-induced environment shall be applied in accordance with [KSC-STD-164](#).

4.2.4.2.1 Launch-Induced Damage

GS not required to function after exposure to the launch-induced environment shall not cause damage or create a hazard to flight hardware, facilities, other GS, personnel, or the environment.

4.2.4.3 Controlled Interior Environment

GS designed to function within a controlled interior environment shall be designed to the following temperature and humidity requirements:

- a. Temperature: +15 °C (60 °F) to +27 °C (80 °F) and within the extremes of +11 °C (52 °F) to +40 °C (104 °F) for a maximum of 1 hour.
- b. Humidity: nominal 55%, within a range of 30% to 70%.

4.2.4.4 Controlled Clean Environment

- a. GS shall be designed to meet the program/project contamination control requirements.
- b. Clean rooms and associated controlled environments designated as GS shall be designed in accordance with [ASTM E2217](#).

4.2.4.5 Uncontrolled Interior Environment

GS used in an uncontrolled interior environment shall be designed to meet the most severe exterior environmental conditions for humidity and temperature expected at the respective geographical locations, as defined in [NASA-HDBK-1001](#).

Some uncontrolled interior environments can exceed the most severe exterior environment, e.g., an enclosed trailer in a hot (tropical or desert) exterior environment.

4.2.4.6 Fire/Explosion Hazardproofing

If GS will be operated in locations where fire or explosion hazards exist, as defined by [NFPA 70](#), Article 500, it shall be hazardproofed in accordance with the requirements in [KSC-STD-E-0002](#).

4.2.4.7 Environmental Test Methods

Environmental methods and conditions required for testing and qualification of GS components shall be in accordance with [MIL-STD-810](#) and [KSC-STD-164](#).

4.2.4.8 Seismic Environment

GS used in Zones 3 or 4 as defined in [NASA-HDBK-1001](#) shall be designed to resist the effects of a seismic event using the criteria and guidelines in [ASCE-7](#).

4.3 Documentation

Documentation in [Appendix B](#) should be released and made available to the user.

4.3.1 Drawings and Specifications

Drawings and specifications required for the fabrication, construction, installation, modification, test, operation, maintenance, sustaining, and use of GS shall be prepared in accordance with drawing practices equal to or more stringent than the engineering drawing practices of [ASME Y14.100](#) and [KSC-GP-435](#), Vol. I.

4.3.2 Technical Documentation

Technical documentation (e.g., manuals and reports) shall be prepared in accordance with [KSC-DF-107](#).

4.4 Logistics

GS design documentation shall identify spare parts, components, materials, and items necessary to support construction, fabrication, installation, activation, test, verification, and operation.

4.4.1 Limited-Life Item Tracking

- a. Limited-life items shall be controlled from the date of their manufacture through their period of operational use, including the time they are in storage.
- b. The status of limited-life items shall be documented.
- c. GS design documentation shall identify limited-life items and provide instructions for replacement of those parts, components, and materials.

4.5 Qualification

All components used in critical GS shall undergo qualification to verify performance in their intended environment in accordance with [KSC-STD-G-0003](#).

Qualification of components is guided by [NPD 8730.2](#).

4.6 Quality Assurance

Quality assurance is the function that verifies the compliance of hardware and software to specified requirements. Quality assurance includes quality engineering and quality inspection. KSC minimum requirements and best practices are given in [KNPR 8730.2](#).

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4.6.1 General

GS design shall incorporate quality requirements in accordance with the program's/project's S&MA Plan.

Quality assurance best practices for applying program requirements at KSC are given in [KSC-NE-9439](#), Section 5.4.

4.6.1.1 Definition of Quality Requirements

Quality requirements shall be defined on the engineering drawings or in other technical documents that are included in the design, fabrication, or installation contract.

The design documentation should include special quality-related requirements, such as any special processes, certification of personnel or special testing that should be conducted, and any other special requirements that are necessary to produce a quality product.

4.6.1.2 Personnel Training

Certification documentation for required training of personnel performing processes during fabrication, assembly, installation, and testing of GS shall be maintained in accordance with [KNPR 8730.2](#).

4.6.2 Testing

- a. Testing requirements shall be specified in engineering documentation.
- b. Testing shall verify compliance with the applicable specifications and the ability of the GS to perform its required design functions.
- c. Test documentation, data, and results shall be produced, maintained, and archived in accordance with [KNPR 8730.2](#) and program or project requirements.

4.6.2.1 Load Test

- a. A load test shall be performed on all structural GSE (access platforms, workstands, etc.) and moveable or articulating GS structures except when the load test creates an unsafe condition or when analysis shows the load test will result in damage to the structure.

Load testing of fixed GS structures may be accomplished using a representative section of the structure.

- b. The minimum test load shall be 125% of the design or working load for GS not used for lifting.
- c. Lifting devices and equipment shall be proofload tested in accordance with [NASA-STD-8719.9](#).

- d. In the absence of design analysis, a load test of 8 times the safe working load shall satisfy the factor of safety requirements of this standards manual.

This load test is to eliminate the need for structural analysis. The 8-to-1 load test screens out flaws that could propagate to failure in a structure used at or below the design load. This option should only be used for ductile materials.

4.6.2.2 Nondestructive Evaluation (NDE)

NDE for components and materials shall be in accordance with [6.4.15](#).

4.6.2.3 Metrology and Calibration

Metrology and calibration of all test equipment and tools used in support of fabrication, assembly, installation, and test of GS shall be in accordance with [KNPR 8730.1](#).

4.6.3 Quality Conformance Verification

Documentation shall be provided by the design organization to verify compliance with this standard in accordance with the program/project verification plan (see [Appendix B](#)).

Test, analysis, demonstration, and inspection (including similarity) are recommended methods to verify that each requirement of Sections 4 and 5 of this standard has been satisfied.

4.7 Packaging, Handling, and Transportation

Requirements for packaging, transporting, shipping, and handling shall be in accordance with [NPR 6000.1](#).

GS should be designed so it can be transported by ground, air, or sea, using commercially available methods.

4.7.1 Shipping and Storage Containers

- a. Containers shall be compatible with onsite transportation, handling, and storage methods.
- b. Container attachment points shall be provided for crane hoists and tie-downs.
- c. Containers shall be designed so that indicators that require monitoring (e.g., desiccants, humidity monitors, shock meters, and tilt meters) can be monitored without opening the shipping container.
- d. Containers shall be marked in accordance with [NPR 6000.1](#) so that contents can be identified without opening the container.

Containers having a gross weight of more than 65 kilograms (144 pounds) should be provided with integral skids or pallets for shipment.

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4.7.2 Parts Protection

- a. Procedures shall be employed to protect parts during manufacturing and in-plant handling and storage.

Any procedures, methods, materials, and devices (such as carts, boxes, containers, or transportation vehicles) that are used to protect parts should be standardized to prevent damage to hardware.

- b. Precision-cleaned parts shall be packaged in accordance with [NPR 6000.1](#) and program contamination control requirements.

5. DESIGN AND CONSTRUCTION REQUIREMENTS

5.1 Structural Design

5.1.1 GS Structures and Equipment

5.1.1.1 Steel Structures

- a. The design of conventional steel structures shall be in accordance with [AISC 325](#).
- b. The design of nonconventional steel structures shall be in accordance with established design practices using the safety factors in [5.1.2](#).

Nonconventional steel structures may be designed using provisions of [AISC 325](#), provided that the factors of safety are adjusted to be consistent with [5.1.2](#).

- c. High-strength carbon steel fasteners such as [ASTM A490](#) and SAE Grade 8 shall not be used in a corrosive environment.

Galvanizing and other zinc coatings may cause hydrogen embrittlement in high-strength carbon steel fasteners that is difficult to detect. When possible, [ASTM A325](#) or SAE Grade 5 fasteners should be used instead.

5.1.1.2 Aluminum Structures

- a. The design of conventional aluminum structures shall be in accordance with [ADM](#).
- b. The design of nonconventional aluminum structures shall be in accordance with established design practices using the safety factors in [5.1.2](#).

Nonconventional aluminum structures may be designed using provisions of [ADM](#), provided that the factors of safety are adjusted to be consistent with [5.1.2](#).

5.1.2 Safety Factor

- a. The safety factor shall not be used to justify exceeding the safe working load.

- b. The following minimum safety factors shall be used for support structures (excluding lifting devices, pressure vessels, threaded fasteners, and springs) when not otherwise specified.
- (1) GS structures shall be designed to a minimum safety factor of 2 against deformation or yielding that impairs the function of the part and a minimum safety factor of 3 against collapsing, buckling, exceeding the ultimate load, or failing to support the design load.
 - (2) For brittle metals (less than 5% elongation-to-failure), excluding castings, the factor of safety shall be 5 against exceeding the ultimate tensile strength (UTS).
 - (3) For metal castings, the factor of safety shall be 10 against exceeding the UTS.
 - (4) For ductile-matrix composites with test data, the factor of safety shall be 5 against exceeding the UTS; in the absence of test data, the factor of safety shall be 10.
 - (5) For brittle-matrix composites, the factor of safety shall be 10 against exceeding the UTS.

Due to the nature of their design, springs do not meet the factors of safety specified for other engineering applications and should be designed in accordance with the Handbook of Spring Design from the Spring Manufacturers Institute.

5.1.2.1 Safety Factor for Cyclic Loading

Structures exposed to cyclic loads shall be designed for a minimum safety factor of 4 against the design life.

The ratio of fatigue life to design life is 4 as a minimum. The designer/analyst has the responsibility to determine when cyclic loads are significant in the design. The relationship between stress/strength and fatigue life is highly nonlinear. Applying the safety factor to life is the established engineering method to handle this nonlinearity. The structure is designed to withstand four times the number of load cycles expected during the design life.

5.1.3 Structural Design Loads

Structural design loads shall be specified in the design documentation.

The design should consider loads created by the assembly, lifting, handling, transportation, operations, wind conditions, launch-induced environments, and seismic events. For torque loads on threaded fasteners, see [5.2.9](#).

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5.2 Mechanical Design

5.2.1 Pneumatics

The design of GS used for pneumatic servicing shall be in accordance with [KSC-STD-Z-0005](#).

Media includes but is not limited to nitrogen, helium, hydrogen, oxygen, methane, breathing air, and special mixtures of these gases.

5.2.1.1 Breathing-Air Systems

- a. Breathing-air respirator systems shall conform to [29 CFR 1910](#).
- b. Breathing-air storage and distribution systems shall be designed in accordance with [KSC-STD-Z-0008](#).

5.2.1.2 Vacuum and Compressed-Air Systems

- a. Vacuum systems shall be designed in accordance with [ASME B31.9](#).
- b. Compressed-air systems having an operating gage pressure up to 1.0 megapascals (MPa) (150 pounds per square inch [psi]) shall be designed in accordance with [ASME B31.9](#).
- c. Compressed-air systems above 1.0 MPa (150 psi) shall be designed in accordance with [ASME B31.3](#) and [KSC-STD-Z-0005](#).

[ASME B31.1](#) meets or exceeds the requirements of [ASME B31.3](#) and may be used as an alternative.

5.2.2 Cryogenics

- a. The design of GS used for cryogenic servicing with liquid hydrogen, liquid oxygen, and liquid nitrogen shall be in accordance with [KSC-STD-Z-0009](#).
- b. The design of GS used for cryogenic servicing with liquid methane shall be in accordance with [NFPA 59A](#).

The design of GS used for cryogenic servicing with liquid helium and cryogenic fluids should use [KSC-STD-Z-0009](#) as a guide.

5.2.3 Hypergols

The design of GS used for hypergolic-fuel or oxidizer servicing with monomethylhydrazine (MMH), unsymmetrical dimethylhydrazine (UDMH), nitrogen tetroxide (N₂O₄), Aerozine 50 (A-50), or hydrazine (N₂H₄) shall be in accordance with [KSC-STD-Z-0006](#).

Hypergolic propellants are fuels and oxidizers or fuels and catalysts that, when combined, produce a violent, explosive, high-energy exothermic reaction, without any other ignition source. Hypergolic propellants are highly toxic, are very sensitive to material selection for containment or sealing, and require the use of personal protective equipment (PPE) for operators of GS.

5.2.4 Hydrocarbons

The design of GS used for servicing with hydrocarbon fuels (JP-4, JP-5, RP-1, and ASTM jet fuels A and B) shall be in accordance with [ASME B31.3](#).

ASME B31.1 meets or exceeds the requirements of [ASME B31.3](#) and may be used as an alternative.

5.2.5 Hydraulics

The design of GS used for servicing hydraulic systems shall be in accordance with [ASME B31.3](#) and with [KSC-STD-Z-0005](#).

ASME B31.1 meets or exceeds the requirements of [ASME B31.3](#) and may be used as an alternative.

5.2.6 Environmental Control Systems (ECSs) and Environmental Control and Life Support Systems (ECLSSs)

- a. The design of GS used for an ECS or ECLSS shall be in accordance with [KSC-STD-Z-0010](#).
- b. The design of ducting shall be in accordance with [ASME B31.3](#) or [SMACNA 1958](#).

ASME B31.1, Power Piping, meets or exceeds the requirements of [ASME B31.3](#) and may be used as an alternative.

5.2.7 Life Support

The design of GS used for life support systems shall be in accordance with [KSC-STD-Z-0008](#).

GS intended for use by operators wearing propellant handler's ensembles (PHEs) should be designed to meet the following criteria:

- a. *Items (valves, gages, levers, bolts, nuts, and any other items required to be moved, turned, manipulated, or monitored) should be located in a position that will make it easier for a PHE-suited operator to access the item while standing.*
- b. *Sufficient clearance should be provided to preclude the operator from brushing against other surfaces.*

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- c. *GS should be designed to avoid requirements for PHE-suited operators to reach into tight areas; stoop to avoid low overhead obstructions; mount supplementary ladders or stairs; touch rough surfaces; or sit, kneel, or lie on the floors or decks.*
- d. *The design should include suitable provisions to prevent discomfort or fatigue for the PHE-suited personnel.*
- e. *Use of expanded metal surfaces and other sharp edges should be avoided.*

5.2.8 Lifting Devices

The design and certification of lifting devices (cranes, crane girders, hoists, lifting slings, jacks, etc.) shall be in accordance with [NASA-STD-8719.9](#).

5.2.9 Torque for Threaded Fasteners

- a. The combined stress from all applied loads (plus an uncertainty factor), including the preload, shear, tension, and bending stress, shall not exceed the yield strength of the fastener material.
- b. The torque limit for threaded fasteners that are less than or equal to 32 millimeters (mm) (1 1/4 inches) in diameter shall be in accordance with [MSFC-STD-486](#) or the alternative limit identified on the drawing.
- c. Threaded fasteners greater than 32 mm (1 1/4 inches) in diameter shall be designed for a maximum torque that results in a preload of 70% of the yield stress on the net cross section.
- d. The criteria for torquing structural bolts, such as specified in [ASTM A325](#) and [ASTM A490](#) shall be in accordance with [AISC 325](#).

[ASTM A490](#) structural bolts should be avoided in a corrosive environment. Galvanizing and other zinc coatings may cause hydrogen embrittlement that is difficult to detect. When possible, [ASTM A325](#) bolts should be used instead.

- e. Torquing criteria shall be documented on the fabrication, assembly, or installation drawing.

For nonstructural applications where a specific clamping force is not required, the designer may specify that fasteners be installed snug tight (tighten the fastener with standard tools, using ordinary force, until the assembly layers come into firm contact, determined visually and by feel).

5.2.10 Tethering Provisions

- a. GS components that require temporary removal/installation during operations, such as quick-release pins and quick-disconnect (QD) caps, shall be tethered to or otherwise held captive by the equipment for which they are used.
- b. GS intended for use near flight hardware or elevated above personnel or flight hardware shall be designed and constructed with provisions for tethering.
- c. Quick-release pins and pin tethers shall be in accordance with [KSC-STD-P-0006](#).

5.2.11 Jacks

The design of jacks shall be in accordance with [ASME B30.1](#).

5.2.12 Transportation Equipment

GS used for transporting flight hardware shall be designed in accordance with [SAE ARP 1247](#).

5.2.12.1 Towed GS

Towed GS shall be designed in accordance with [SAE AS 8090](#).

5.2.12.2 Transportation Equipment Interface Loads

Transportation equipment shall be designed so that loads imparted to flight hardware do not exceed 80% of the flight limit loads.

Transportation loads should be evaluated early in the design cycle since they may be the governing design load case.

5.2.13 Pressure Vessels

- a. Metallic pressure vessels for use in GS shall be designed, constructed, tested, and stamped in accordance with [ASME BPVC-VIII](#), Division 1, 2, or 3.

Shops and welders repairing or altering code vessels require R stamp certification. For noncode vessels, R stamp certification is not required but is good practice.

- b. Composite Overwrapped Pressure Vessels (COPVs) for use in GS shall be designed, constructed, tested, and marked in accordance with [ASME BPVC-X](#).

COPVs should be used only when necessary because of weight constraints. Special training and certification are required for personnel handling and repairing COPVs. Damage to COPVs can be difficult to detect because reliable NDE methods have not been developed. COPV failure modes can be catastrophic, requiring special safety clears during pressurization.

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5.2.13.1 Code-Stamped Vessel Registration

All ASME code-stamped vessels shall be registered with the National Board of Boiler and Pressure Vessel Inspectors.

5.2.13.2 Pressure Vessels Used in Transporting Commodities

Pressure vessels used for transporting hazardous commodities shall meet the Department of Transportation requirements in [49 CFR 171 through 180](#).

5.2.14 Piping Systems

Piping and support systems, except those in remote or “cross country” areas or where personnel traffic is low, shall be in accordance with [ASME B31.3](#) the piping system may be designed to [ASME B31.8](#).

Examples include piping between facilities in low-traffic areas, buried pipelines, and long-run piping between remote facilities. Where [ASME B31.3](#) piping connects to [ASME B31.8](#) piping e.g., at facility interfaces), the design should consider the differences in internal diameters and how the piping will be cleaned (via pipeline cleaning pigs). [ASME B31.1](#) meets or exceeds the requirements of [ASME B31.3](#) and may be used as an alternative.

5.2.15 Pressure Vessel/System Certification

The design of pressure systems, including pressure vessels, transmission lines, and GS, shall comply with [KNPR 8715.3](#), Section 13, which complies with [NASA-STD-8719.17](#).

5.2.16 Ground-to-Flight-Vehicle Umbilical Systems

Reusable umbilicals shall be designed in accordance with [KSC-GP-986](#).

5.3 Electrical/Electronic Design

5.3.1 Electrical Control and Monitor Equipment

The design of GS for electrical control and monitoring shall be in accordance with [KSC-STD-E-0001](#).

[AFSPCMAN 91-710](#) and [MIL-HDBK-454](#) should be consulted for general guidance in electrical/electronic design.

5.3.2 Electrical Design of Pneumatic and Hydraulic Components

The electrical design of pneumatic and hydraulic components shall be in accordance with [KSC-STD-E-0004](#).

5.3.3 Pyrotechnic Systems

The design of pyrotechnic GS shall be in accordance with [NASA-STD-8719.12](#).

5.3.4 Electrical Power Systems

The design of electrical power systems covered by [NFPA 70](#) shall be in accordance with [NFPA 70](#) and [NFPA 70E](#).

Incorporation of batteries in the design of GS should follow the recommended practices in the following documents: [IEEE 484](#), [IEEE 1106](#), [IEEE 1187](#), [IEEE 446](#), [ANSI C18.2M, Part 1](#), and [ANSI C18.3M, Part 1](#).

5.3.5 Bonding and Grounding

- a. Bonding and grounding shall be provided in accordance with [NFPA 70](#) and [NASA-STD-4003](#).
- b. GS shall be designed in accordance with [KSC-STD-E-0012](#), [KSC-E-165](#), and [MIL-STD-464](#).

The designer should be aware whether the system being designed will be operated under a lightning protection system or where induced static electricity is possible. For additional information on lightning protection, refer to International Electrotechnical Commission standard [IEC 62305](#), Parts 1 through 4.

5.3.6 Hazardproofing

The design of electronic equipment and wiring for all voltages in hazardous locations shall be in accordance with [NFPA 70](#), Article 500; and [KSC-STD-E-0002](#).

5.3.7 Software

Software incorporated into the design of GS shall meet the requirements of [NPR 7150.2](#).

This ensures that KSC-developed software meets the Agency requirements for software engineering practices, software assurance, and software safety. [NPR 7150.2](#) contains provisions applicable to COTS software in NASA-developed systems.

5.3.8 Firmware

Firmware incorporated in the design of GS shall meet the requirements of [NPR 7150.2](#).

5.4 Parts

5.4.1 Electrical, Electronic, and Electromechanical (EEE) Parts

EEE parts shall be selected in accordance with [KNPR 8720.2](#).

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EEE parts may be selected using [AIAA R-100](#) or [NASA/TP-2003-212242](#).

5.4.1.1 Electrostatic-Discharge (ESD)-Sensitive Components and Assemblies

All ESD-sensitive components and assemblies shall be handled using practices in accordance with [ESD S20.20](#), [MIL-STD-1686](#), and [MIL-HDBK-263](#).

Insulator materials should not be used near ESD-sensitive components or assemblies.

5.4.2 Stainless-Steel Tubing

- a. Austenitic stainless-steel tubing shall be in accordance with [KSC-SPEC-Z-0007](#).
- b. Stainless-steel tubing shall be fabricated, tested, and installed in accordance with [KSC-SPEC-Z-0008](#).

5.4.2.1 Superaustenitic Stainless-Steel Tubing

- a. When directly exposed to a corrosive environment, tubing shall consist of Unified Numbering System (UNS) N08367 or S31245 superaustenitic stainless steel (trade name AL6XN or 254SMO) in accordance with [KSC-SPEC-P-0027](#).
- b. Superaustenitic stainless-steel tubing shall be fabricated, tested, and installed in accordance with [KSC-SPEC-Z-0008](#).

5.4.3 Pipe

Stainless-steel pipe for fluid systems shall be in accordance with [ASTM A312](#).

5.4.3.1 Aluminum Pipe

Aluminum pipe for fluid systems shall be in accordance with [ASTM B241](#).

5.4.3.2 Expansion Joints

Expansion joints used in fluid systems in marine or launch-induced environments shall be made from UNS N06022 (Hastelloy C22) material.

5.4.4 Metallic Fittings

Flared tubing fittings, tube weld fittings, and pipe fittings shall be selected in accordance with [KSC-GP-425](#).

5.4.5 Fluid System Protective Covers

Protective covers shall be provided for all hoses, ports, fittings, and other fluid-fitting connections to GS to protect the threads, protect the sealing surface, and maintain the cleanliness

of the system. Prior to installation, protective caps, plugs, and covers shall be maintained clean at the same level or better than the system that they are protecting.

Caution should be used in selecting caps and plugs as covers due to the potential for generating debris during installation or removal, especially in oxygen systems. When possible, the protective cover should be connected with a lanyard or the equipment should have a designated storage provision. Caps and plugs should be selected in accordance with [KSC-C-123](#).

5.4.6 Fluid System Components

Fluid system components used in the design of liquid or gas systems shall be selected from the [79K80000](#) series of specifications, unless the required part is not covered by this series.

5.4.6.1 Fluid System Component Acceptance Criteria

If a part is not covered by the [79K80000](#) series of specifications, it shall be documented with the following minimum information: commodity, environment, performance, installed dimensions, connection interfaces, recommended vendor, materials, compatibility, qualification/acceptance criteria, and recommended maintenance.

5.4.7 Electrical-Power Receptacles and Plugs

- a. Alternating-current (AC) electrical-power receptacles and plugs that supply power to GS shall be in accordance with [KSC-STD-E-0011](#).
- b. Electrical-power receptacles and plugs within GS shall be in accordance with [KSC-GP-864](#).

5.4.8 Electrical-Power Cable

Power cables shall be in accordance with [NFPA 70](#).

5.4.9 Electrical Cable and Harnesses

Flexible, multiconductor, jacketed electrical cable and cable harnesses shall be in accordance with [KSC-GP-864](#).

5.4.10 Fiber Optics

- a. Fiber-optic cable assemblies, installations, and terminations shall be in accordance with [NASA-STD-8739.5](#) and [KSC-GP-864](#).
- b. Protective caps shall be provided for all fiber-optic connections to GS so that the mating surface is protected.

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5.4.11 Electrical Hookup Wire

Electrical hookup wire shall be in accordance with [SAE AS 50861](#), [MIL-DTL-16878](#), or [SAE AS 22759](#).

5.4.12 Connectors

- a. Electrical multiconductor connectors for GS used for electrical control and monitoring shall be selected from the following documents: [SAE AS 50151](#), [MIL-DTL-22992](#), [MIL-DTL-24308](#), [MIL-DTL-38999](#), [IEC 60807](#), or [KSC-GP-864](#).

MIL-DTL-5015 was replaced with [SAE AS 50151](#).

- b. When multiconductor connectors must be used in hazardous locations, they shall be waterproof and used with threaded connectors.

5.4.13 Coaxial Radio Frequency (RF) Connectors

Coaxial RF connectors shall be selected from [MIL-PRF-39012](#).

5.4.14 Protective Covers or Caps for Electrical Connectors

Protective covers or caps shall be specified for use with electrical-connector plugs and receptacles in accordance with [KSC-GP-864](#) when they are not connected.

Metallic protection caps should be used during transportation or other potentially damaging handling situations.

Protective covers or caps should

- *protect against moisture intrusion,*
- *protect sealing surfaces, threads, and pins against damage,*
- *resist abrasion, chipping, and flaking,*
- *comply with cleanliness requirements for the plugs and receptacles on which they are used,*
- *consist of material that is compatible with the connector materials,*
- *connect to the cable with a suitable lanyard, chain, or hinge, and*
- *not produce static.*

5.4.15 Optical Covers or Caps

Covers/caps shall be provided to protect optics.

Optical covers/caps should be easy to install and remove. The covers/caps should be connected with a lanyard, or the equipment should have a designated storage provision.

5.4.16 Sensors and Transducers

Sensors and transducers used in the design of GS systems shall be selected using [KSC-NE-9187](#).

Measurement applications that provide visibility only and are not relied upon to control a condition that could potentially damage flight hardware or potentially create a safety hazard may use COTS components.

5.4.16.1 Sensor and Transducer Acceptance Criteria

If a part is not covered by [KSC-NE-9187](#), it shall be documented with the following minimum information included: commodity, environment, performance, recommended vendor, materials, compatibility, and qualification/acceptance criteria.

5.4.17 Purged Electrical Enclosures

Purged electrical enclosures in hazardous locations shall be in accordance with [NFPA 496](#). Purged electrical enclosures in nonhazardous locations shall be outfitted with purge hardware in accordance with [79K07491](#).

[NFPA 496](#) contains requirements for the protection of electrical and electronic equipment recognized by [NFPA 70](#) for installations in hazardous locations. Hardware required for purging enclosures (to maintain low relative humidity and positive pressure) should be selected from [79K07491](#).

5.4.18 Racks, Panels, and Modular Enclosures

Electronic racks, panels, and enclosures shall be in accordance with [KSC-SPEC-E-0002](#).

Electronic racks, panels, and modular enclosures should conform to the configuration and dimensional requirements of [ECA EIA/ECA 310-E](#).

5.4.19 Printed Circuit Boards

5.4.19.1 Printed Circuit Board Design

Rigid, flexible, and rigid-flex printed circuit boards (single, double, metal-core, or multilayer structures) shall meet the design specifications of the following standards, as applicable: [IPC-2221](#) (For critical applications, performance classification 3 shall be used.), [IPC-2222](#), [IPC-2223](#), and [IPC-2252](#).

All board types within the [IPC-2221](#) series standard documents are acceptable.

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5.4.19.2 Printed Circuit Board Fabrication and Acceptance

Rigid, flexible, and rigid-flex printed circuit boards (single, double, metal-core, or multilayer structures) shall meet the qualification and performance specifications of the following standards, as applicable: [IPC-6011](#), (For critical applications, performance classification 3 shall be used.), [IPC-6012](#), (For critical applications, performance classification 3 shall be used.), [IPC-6013](#), and [IPC-6018](#).

5.4.19.3 Printed Circuit Assembly Fabrication and Acceptance

- a. Printed circuit assemblies that will not be exposed to vibration or thermal cycling at space flight levels shall be fabricated in accordance with [IPC J-STD-001](#).

[NASA-STD-8739.2](#) and [NASA-STD-8739.3](#) may be used in lieu of [IPC J-STD-001](#).

- b. Printed circuit assemblies that will be exposed to vibration or thermal cycling at space flight levels shall be fabricated in accordance with [IPC J-STD-001ES](#).

[NASA-STD-8739.2](#) and [NASA-STD-8739.3](#) may be used in lieu of [IPC J-STD-001ES](#).

5.4.20 Electric Motors and Generators

Motors and generators used in GS shall be selected in accordance with the system requirements for speed, torque, horsepower, and environment. Motors and generators shall meet the applicable National Electrical Manufacturers Association (NEMA) standards and NFPA requirements that govern the classification and general application of motors and generators.

[NEMA MG 1](#) is a comprehensive document that includes the classification, general standards, manufacturing, and test of motors and generators. This document should be used as a guide for specification and selection of GS motors and generators. Another reference and guide is [NEMA C50.41](#). [NFPA 70](#) provides the size of wiring and conduit required for the motor connections.

5.4.20.1 Motor Starters and Controllers

Motor starters and controllers shall be in accordance with the type of motor, performance ratings, and type of control required. Enclosures for motor starters and controllers shall meet the environmental requirements for their locations.

[NEMA ICS 2](#), Parts 1 through 9, is a comprehensive document that provides practical information concerning ratings, construction, test, performance, and manufacture of industrial control equipment. [NEMA ICS 61800-2](#) specifies ratings for low-voltage, adjustable-frequency AC power drive systems. [NFPA 70](#) provides the size of wiring and conduit required for the starter/controller connections.

5.4.21 GS Fasteners

GS fasteners used in critical applications shall have lot traceability from the manufacturer to final installation.

NASA RP-1228 should be used for guidance in selecting and analyzing fasteners.

5.4.21.1 Reuse of Self-Locking Fasteners

- a. The reuse of self-locking fasteners shall be permitted when the running torque before clamp-up remains between the maximum self-locking torque and the minimum break-away torque.

Self-locking fasteners should be used wherever possible. The use of star lock washers should be avoided.

- b. Fasteners used in corrosive environments and applications where condensation can occur shall be installed using a corrosion-resistant sealant while the sealant is still wet (wet installation).

5.4.21.2 Liquid-Locking Compounds

When liquid-locking compounds are used for fastener installation, engineering drawings shall specify a validated process for application.

Liquid-locking compounds should be selected in accordance with [ASTM D5363](#).

5.5 Electromagnetic Interference (EMI)

Electrical and electronic systems shall be designed to meet the EMI/electromagnetic-compatibility (EMC) requirements in [MIL-STD-461](#).

The requirements of [MIL-STD-461](#) may be tailored for use on NASA programs as indicated in [MIL-STD-461](#), Sections 1.2, 1.2.1, and 1.2.3. The application of [MIL-STD-461](#) to GS should be based on an evaluation of the potential for flight hardware interaction and any existing commercial standards to which the hardware is already certified.

5.6 Identification Markings and Labels

5.6.1 Systems and Equipment

- a. GS shall be identified and marked in accordance with KSC-STD-E-0015.
- b. When data matrix identification symbols are used to identify GS, they shall be applied in accordance with [NASA-STD-6002](#).

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5.6.2 Load Test

GS that have been load-tested shall be identified and marked in accordance with [KSC-STD-141](#), and for lifting equipment, [NASA-STD-8719.9](#).

5.6.3 Piping Systems

Ground piping systems shall be identified and color-coded in accordance with [KSC-STD-SF-0004](#).

5.6.4 Compressed-Gas Cylinders

Compressed-gas cylinders shall be labeled in accordance with [CGA C-7](#).

5.6.5 Load Capacity

GS used for transportation, handling, and personnel access shall be conspicuously marked to indicate the maximum safe working load in accordance with [KSC-STD-E-0015](#), and for lifting equipment, [NASA-STD-8719.9](#).

5.6.6 Electrical-Cable and Harness Assemblies

- a. Electrical cable and harness assemblies shall be identified at each end of the cable and/or harness and labeled to show the assembly part number, cable or harness reference designation number, and cable or harness end marking, in accordance with the requirements provided on the cable assembly drawing and described in [120E3100003](#).
- b. Labels for cable assemblies shall comply with [KSC-E-166](#).
- c. Labels shall meet the requirements of [KSC-STD-E-0015](#).
- d. Labels for wire harnesses shall comply with [KSC-E-166](#).
- e. Labels shall be of heat-shrinkable sleeving that conforms to the requirements of [SAE AMS-DTL-23053/5](#), Class 1.
- f. The heat-shrink labels shall be marked in accordance with [SAE AS 5942](#) and shall meet the testing requirements of [MIL-STD-202](#), Method 215.

MIL-DTL-23053/5 was replaced by [SAE AMS-DTL-23053/5](#). MIL-M-81531 was replaced by [SAE AS 5942](#).

5.6.7 Serial Numbers

Serial numbers or other unique identifiers shall be marked on those parts, components, or assemblies that contain limited-life items (e.g., valves or regulators) or that require periodic

inspection, checkout, repair, maintenance, servicing, or calibration (e.g., pressure transducers or gages).

Other unique identifiers include A-numbers and drawing dash numbers for end items.

5.7 Interchangeability

Hardware assemblies, components, and parts with the same part number shall be physically and functionally interchangeable.

5.8 Safety

All GS shall be designed and fabricated in accordance with [29 CFR 1910](#), [29 CFR 1926](#), and [NPR 8715.3](#).

5.8.1 Hazard Analysis

A hazard analysis shall be conducted in accordance with program-defined methodology as part of the GS design process to identify, mitigate, and control hazards. In the absence of a program-defined methodology, [KNPR 8700.2](#) shall be used.

5.8.2 Safety Requirements on KSC Property

GS to be used at KSC shall meet the safety requirements of [KNPR 8715.3](#).

5.8.3 Safety Requirements on Air Force Property

GS to be used on Air Force property shall meet the requirements of [AFSPCMAN 91-710, Volume 3](#).

5.8.4 Safety Requirements on Other NASA Property

GS to be used at other NASA facilities shall meet the safety requirements of those facilities.

5.8.5 Caution and Warning Indications

GS shall provide caution and warning indications to alert personnel of impending or existing hazards.

GS should be designed to allow efficient implementation of the applicable Occupational Safety and Health Administration (OSHA) lockout/tagout requirements.

5.9 Human Factors

- a. [HF-STD-001](#) shall be used to establish human factors criteria for GS design.
- b. GS shall comply with [29 CFR 1910](#).

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5.10 Security of Information Technology (IT)

The design of GS shall meet the IT security requirements in [NPR 2810.1](#).

6. MATERIALS AND PROCESSES REQUIREMENTS

M&P used in the design and fabrication of GS shall be selected by considering the worst-case operational requirements for the particular application and the design engineering properties of the candidate materials.

For example, the operational requirements should consider operational temperature limits, loads, contamination, life expectancy, exposure to moisture or other fluids, corrosive environments, and launch-induced and natural environments. Properties that should be considered in material selection include mechanical properties, fracture toughness, flammability and offgassing characteristics, corrosion and stress corrosion resistance, thermal- and mechanical-fatigue properties, glass-transition temperature, mismatches between coefficients of thermal expansion, vacuum outgassing, fluid compatibility, microbial resistance, moisture resistance, fretting, galling, and susceptibility to ESD and contamination.

6.1 Material Properties Design Data

- a. The following documents shall be used to establish materials properties for use in system or component design: [DOT/FAA/AR-MMPDS](#), [MIL-HDBK-17-2](#), [MIL-HDBK-17-4](#), [MIL-HDBK-17-5](#), and voluntary-consensus standards or codes (e.g., [ASME BPVC-VIII](#) and [ASME BPVC-X](#) for pressure vessels, and [AISC 325](#) for structural steel).

The values listed in the codes or standards are minimum material properties. The use of minimum material properties, as stated by the code, is intrinsic to the factor of safety, margin of safety, strength factor, etc., of the design.

- b. Parts machined from polychlorotrifluoroethylene (PCTFE) shall comply with [ASTM D7194](#).

[ASTM D7194](#) requires an annealing process for parts machined from PCTFE. Annealing results in dimensional stability, which minimizes the occurrence of flow friction in oxygen system components with PCTFE soft goods.

- c. When mechanical properties of new or existing structural materials are not available, they shall be determined by the analytical methods described in the following documents: [MIL-HDBK-17-1](#), [MIL-HDBK-17-3](#), [MIL-HDBK-17-4](#), [MIL-HDBK-17-5](#), [MIL-HDBK-149](#), and [MIL-HDBK-700](#).

If the material is not covered by a design code or one of these sources, the Aerospace Structural Metals Database or other published industry sources should be used in accordance with the best practices for design. The properties listed in these documents are typical values, not minimum values; this must be considered when applying the factor of safety appropriate for the design.

6.2 M&P Controls

M&P controls shall be as follows:

- a. All M&P shall be defined by standards or specifications and shall be identified directly on the appropriate engineering drawing.
 - (1) Engineering drawings shall be reviewed for compliance with all applicable M&P requirements.
 - (2) An M&P assessment shall be done concurrently with the design.
- b. The design drawings shall be signed by an authorized M&P engineer.
- c. Composition and properties of all materials and parts shall be certified as required by the procuring document.
- d. The Materials and Processes Technical Information System (MAPTIS) shall be consulted to obtain material codes and ratings for materials, standard and commercial parts, and components.
- e. For critical GS, Material Usage Agreements (MUAs) shall be submitted for all M&P that are technically acceptable but do not meet the M&P requirements of this standard.
- f. For noncritical GS, MUAs shall be submitted for M&P that are technically acceptable but do not meet the requirements of this standard in regard to flammability, compatibility, or stress corrosion cracking (SCC).

For noncritical GS, the M&P organization's approval on the engineering drawing approves deviations from other M&P requirements of this standard. The use of materials and processes that do not comply with the requirements of this standard may still be acceptable in the actual hardware applications. MUAs are submitted using [KSC Form 21-475](#). See [Appendix A](#).

6.3 Detailed Requirements

6.3.1 Flammability and Compatibility Requirements

Materials shall be tested in accordance with [NASA-STD-6001](#).

6.3.1.1 Flammability Control

- a. Materials used for flammability control shall be nonflammable or self-extinguishing in their use configuration as defined by [NASA-STD-6001](#), Test 1 or Test 10.

Material flammability ratings and tests based on [NASA-STD-6001](#) are available in the MAPTIS database for many materials.

- b. The following materials or methods are also acceptable:

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- (1) Ceramics, metal oxides, and inorganic glasses shall be acceptable without prior testing.

When a material is sufficiently similar (chemically and physically) to a material found to be acceptable by testing in accordance with [NASA-STD-6001](#), this material may be used without additional testing if its use is justified on an approved MUA.

- (2) Materials whose flammability and self-extinguishing properties have been tested in accordance with [NASA-STD-6001](#) under conditions more severe than those encountered in the use environment shall be acceptable without further testing, as in the following examples:
 - (a) Materials used in an environment with an oxygen concentration lower than the test level shall be accepted without further testing (provided that the oxygen partial pressure is not greater than the partial pressure at the test level).
 - (b) Materials used in an environment where the concentration is greater than the test level shall be retested or considered flammable.
 - (c) If a material passes the flammability test on a metal substrate, it shall be used on metal substrates of the same thickness or greater.
 - (d) If the material will be used on a thinner or non-heat-sinking substrate (or on no substrate at all), it shall be retested or considered flammable.

Many situations arise in which flammable materials are used in an acceptable manner without testing, but such uses require mitigation practices and MUA approval. Guidelines for assessment and mitigation of hardware flammability characteristics can be found in [JSC 29353](#).

6.3.1.2 Electrical-Wire Insulation Materials

- a. Electrical-wire insulation materials shall be evaluated for flammability in accordance with [NASA-STD-6001](#), Test 4.
- b. Arc tracking shall be evaluated in accordance with [NASA-STD-6001](#), Test 18.

Arc tracking testing is not required for polytetrafluoroethylene (PTFE), PTFE laminate, ethylene tetrafluoroethylene (ETFE), or silicone-insulated wires because the resistance of these materials to arc tracking has already been established.

6.3.1.3 Fluid Compatibility

For the purposes of this standard, the definition of “hazardous fluids” includes gaseous oxygen, liquid oxygen, fuels, oxidizers, and other fluids that could cause corrosion, chemically or physically degrade materials in the system, or cause an exothermic reaction.

6.3.1.3.1 Fluids Other than Oxygen

- a. Materials exposed to hazardous fluids other than oxygen shall be evaluated or tested for compatibility.

NASA-STD-6001, Test 15, tests materials for short-term exposure to fuels and oxidizers. NASA-STD-6001, Test A.7, tests materials for incidental exposure, such as a splash, to fuels and oxidizers. For many materials, material compatibility ratings and test results based on NASA-STD-6001, Test 15, are available in the MAPTIS database.

- b. Appropriate compatibility tests shall be conducted for materials that are subjected to long-term exposure to fuels, oxidizers, or other hazardous fluids. The test conditions are customer-specified and shall simulate the worst-case use environment that would enhance reactions or degradation of the material or fluid.
- c. Material degradation in long-term tests shall be characterized by posttest analyses of the material and fluid to determine the extent of changes in chemical and physical characteristics, including mechanical properties.

6.3.1.3.2 Oxygen Systems

- a. Liquid oxygen and gaseous oxygen systems shall use materials that are nonflammable in their worst-case use configuration, as defined by [NASA-STD-6001](#), Test 17, for upward flammability in GO₂ (or Test 1 for materials used in oxygen pressures that are less than 350 kPa (50 psia)).

Material flammability ratings and test results based on [NASA-STD-6001](#) are available in the MAPTIS database for many materials. [KTI-5210](#) may be consulted for a summary of oxygen compatibility test results for various materials used in liquid oxygen and gaseous oxygen applications.

- b. When a material in an oxygen system is determined to be flammable by Test 17, an oxygen compatibility assessment shall be conducted in accordance with [NASA-STD-6001](#) and the system safety rationale documented in an MUA.
- c. When the oxygen compatibility assessment shows the risk is above an acceptable level, configurational testing shall be conducted to support the oxygen compatibility assessment.
- d. Configurational testing shall exercise the ignition mechanisms identified by the oxygen compatibility assessment, using test methods described in [ASTM MNL 36](#).
- e. The configurational test method and acceptance criteria shall be reviewed and approved as part of the MUA process described in [6.2](#).

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- f. The as-built configuration shall be verified against the oxygen compatibility assessment to ensure that mitigation methods identified in the report were incorporated into the design and construction of the hardware.
- g. For compressed-air systems and oxygen-enriched systems, the need for an oxygen compatibility assessment shall be addressed on a case-by-case basis.

Compressed-air systems and oxygen-enriched systems are inherently less hazardous than systems containing pure oxygen; the hazard increases with oxygen concentration and pressure.

Guidelines on the design of safe oxygen systems are contained in [ASTM MNL 36](#), [ASTM G88](#), [ASTM G63](#), [ASTM G94](#), and [NASA/TM-2007-213740](#).

- h. Oxygen, oxygen-enriched, and compressed-air system components that operate at pressures above 1.83 MPa (265 psia) shall undergo oxygen compatibility acceptance testing at maximum design pressure for a minimum of 10 cycles before being placed into service. Components shall be retested if the results are invalidated by actions occurring after the test (such as rework, repair, or interfacing with hardware for which the cleanliness level is unknown or uncontrolled).

6.3.1.4 Metals

6.3.1.4.1 Carbon and Low-Alloy Steel

Carbon and low-alloy steels heat-treated to strength levels at or above 1,240 MPa (180 ksi) UTS shall require an MUA due to sensitivity to SCC.

The ductile-to-brittle transition temperature exhibited in steels should be considered when using carbon and low-alloy steels in hardware operating in or exposed to low temperatures while in service. For some alloys, the transition temperature may be as high as the ambient temperature.

6.3.1.4.2 Corrosion-Resistant Steel

- a. Unstabilized austenitic steels shall not be used under conditions where the temperature is above 371 °C (700 °F).

Welding should be performed only on low-carbon, stabilized grades, or superaustenitic grades (i.e., UNS S30403, UNS S31603, UNS S32100, UNS S34700, UNS N08367, and UNS S31254). Welding higher-carbon, unstabilized austenitic stainless steel can result in sensitization in the weld heat-affected zone. Sensitization occurs when carbon precipitates from the solid solution, mainly at the grain boundaries, depleting the chromium in the adjacent matrix, which results in localized intergranular attack when the material is exposed to corrosive service conditions. Caution should be exercised in using martensitic and ferritic stainless steels because these are susceptible to hydrogen embrittlement, corrosion, and stress corrosion. Austenitic stainless steels are susceptible to pitting corrosion and crevice corrosion in a marine environment; some austenitic stainless steels are susceptible to SCC in a marine environment.

- b. Free-machining alloys such as UNS S30300 (type 303 SST) and UNS S30323 (type 303Se SST) shall not be used in corrosive environments.
- c. UNS N08367 (trade name AL-6XN) or UNS S31254 (trade name 254 SMO) shall be used in pressure piping and tubing in lieu of 300-series stainless steel when the piping or tubing is directly exposed to a corrosive environment.
- d. Cleaning, descaling, and passivating of stainless-steel parts, assemblies, equipment, and installed systems shall be in accordance with [ASTM A380](#) and [ASTM A967](#).
- e. When acid cleaning baths are used for steel parts, the parts shall be baked in accordance with [SAE AMS 2759/9](#) to alleviate potential hydrogen embrittlement problems.

Hardware should be designed to avoid fretting or wear of stainless-steel alloys. Lubricants and lubricated coatings should be considered for use with stainless-steel materials in applications where they come into contact through a sliding movement. Stainless-steel alloys (such as Nitronic 60) that resist galling should be considered as alternatives.

6.3.1.4.3 Aluminum

- a. Aluminum alloys used in structural applications shall be resistant to general corrosion, pitting, intergranular corrosion, and SCC.
- b. Alloys in the 5000-series containing more than 3% magnesium shall not be used in applications where the temperature exceeds 66 °C (150 °F), because grain boundary precipitation above this temperature can create stress-corrosion sensitivity.

Hardware made with aluminum alloys should not be loaded through the short transverse grain direction because resistance to SCC is at a minimum in that direction.

6.3.1.4.4 Nickel-Based Alloys

Alloys with high nickel content are susceptible to sulfur embrittlement; therefore, any foreign material that could contain sulfur, such as oils, grease, and cutting lubricants, shall be removed before heat treatment, welding, or high-temperature service.

Some of the precipitation-hardening superalloys are susceptible to depletion of the alloying element at the surface in a high-temperature, oxidizing environment. This effect should be carefully evaluated when a thin sheet is used, since a slight depletion could involve a considerable proportion of the cross section of the material.

6.3.1.4.5 Titanium

- a. Areas subject to fretting or wear shall be anodized in accordance with [SAE AMS 2488](#) or hard-coated using a wear-resistant material such as a tungsten carbide/cobalt thermal spray.

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Titanium and its alloys exhibit very poor resistance to wear. Fretting may cause cracking, potentially leading to fatigue failure. The design should avoid fretting or wear in titanium and its alloys.

- b. Titanium alloys shall not be used with LO₂ or GO₂ at any pressure or with air at oxygen partial pressures above 35 kPa (5 psia).
- c. The surfaces of titanium and titanium alloy mill products shall be 100% machined, chemically milled, or pickled to a sufficient depth to remove all oxygen-contaminated layers (alpha case) formed as a result of mill processing, heat treating, and forming operations at elevated temperatures.
- d. All cleaning fluids and other chemicals used during manufacturing and processing of titanium hardware shall be verified to be compatible with and not detrimental to the material's performance.

The use of titanium in hydrochloric acid, chlorinated solvents, chlorinated cutting fluids, fluorinated hydrocarbons, and anhydrous methyl alcohol should be avoided due to titanium's susceptibility to SCC. Contact of titanium alloys with mercury, cadmium, silver, and gold should be avoided at certain temperature ranges because of liquid-metal-induced embrittlement and/or solid-metal-induced embrittlement.

6.3.1.4.6 Copper Alloys

- a. Copper alloys, brasses, and bronzes that are resistant to general corrosion and pitting, and highly resistant to SCC in accordance with [MSFC-STD-3029](#) shall be used in GS.
- b. Copper alloys shall not be used in solutions with ammonium ions or in contact with ammonia to prevent SCC.

GS should be designed so that copper is not exposed to hydrazine environments. Copper has the potential for SCC when exposed to ammonia, which is a product of hydrazine decomposition. Beryllium copper (UNS C17200) is commonly used for high-strength, nonsparking structural components in applications where it is subject to contact and wear.

6.3.1.4.7 Beryllium and Beryllium Alloys

Beryllium particles, beryllium oxide, and other beryllium compounds are toxic when inhaled. Extreme caution must be exercised during fabrication to avoid exposing personnel to beryllium or beryllium compounds.

Machining, grinding, and finishing operations on beryllium and beryllium alloys shall be performed either wet, using a liquid coolant with local ventilation, or dry, using high-velocity, close-capture ventilation.

Refer to the appropriate Material Safety Data Sheet for more detail.

6.3.1.4.8 Tin

- a. Tin solder and tin plating shall be alloyed with at least 3% lead to prevent the growth of tin whiskers.
- b. For critical GS, lot sampling shall be used to verify the presence of at least 3% lead.

6.3.1.5 Nonmetals**6.3.1.5.1 Elastomers**

- a. Elastomers shall be in accordance with [MIL-HDBK-149](#).
- b. Elastomers shall operate within the parameters of a design service life, including the vendor-specified shelf life.
- c. Elastomers shall be cure-dated for tracking purposes.
- d. Elastomers shall not have a corrosive effect on other materials when exposed to conditions normally encountered in service.

Examples include one-part silicones that liberate acetic acid when they are cured.

- e. When rubbers or elastomers are used at low temperatures, the ability of these materials to maintain the required elastomeric properties shall be verified by testing them at or below use temperature.

6.3.1.5.2 Composite Materials

- a. Composite materials used in GS shall be in accordance with MIL-HDBK-17, Volumes 1 through 5.
- b. Defects resulting from manufacturing shall be assessed through NDE techniques to meet the intent of [6.4.15](#).

6.3.1.5.3 Refractory Concrete

Refractory concrete used for heat and blast protection of flame deflectors and other areas of the launch pad shall be in accordance with [KSC-SPEC-P-0012](#).

6.3.1.5.4 Lubricants

[NASA-TM-86556](#), Parts A and B, should be used in evaluating and selecting solid and liquid lubricants for GS. Guidelines on additional lubricants are contained in [NASA/CR-2005-213424](#). Long-life performance should be considered when selecting lubricants. Use of lubricants near precision-cleaned hardware or electrical connections should be minimized or tightly controlled to prevent cross-contamination.

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- a. Lubricants containing chloro-fluoro compounds shall not be used with aluminum or magnesium.
- b. Lubricants containing chloro-fluoro compounds shall not be heated above their maximum temperature rating.

The decomposition/reaction products that result from overheating lubricants that contain chloro-fluoro compounds can attack metallic materials and can be toxic.

- c. Lubrication of flared tube fittings shall be in accordance with [KSC-SPEC-Z-0009](#).

6.3.1.5.5 Limited-Life Materials

Materials that are not expected to meet the design life requirements but must be used for functional reasons shall be identified as limited-life items in accordance with [4.2.2.2](#).

Materials should be selected to meet the useful life of the hardware without the need for additional maintenance. Useful life includes storage life, installed life in a nonoperating mode, and operational service life.

6.3.1.5.6 Plastic Film, Foam, and Adhesive Tape (PFA)

Thin plastic films and tape materials used in GS shall be tested in accordance with and meet the requirements of the following for flammability, ESD, and hypergolic ignition/breakthrough characteristics, respectively, as appropriate for the application: [NASA-STD-6001](#) (Test 1 and A.7) and [MMA-1985-79](#).

Material flammability ratings and hypergol compatibility test results for many PFAs are found in the MAPTIS database. [KTI-5212](#) may be consulted for a summary of flammability, ESD, and hypergol compatibility test results for various PFAs.

6.3.1.5.7 Fungus Resistance

For GS used in a humid environment, materials shall meet one of the following criteria or be identified in the design documentation along with any action required, such as periodic inspection, maintenance, or replacement of the material:

- a. Materials are selected from fungus-inert materials (Group I) identified in [MIL-STD-810](#). Fluorocarbon polymers (including ETFE) and silicones are considered acceptable.
- b. Materials are used under one of the following conditions:
 - (1) Materials are used inside environmentally sealed containers with less than 60% relative humidity (RH) at ambient conditions.

- (2) Materials are used inside electrical boxes where the temperature is always greater than or equal to the ambient cabin temperature.
- (3) Only the edges of materials are exposed.
- c. Materials are tested for fungus resistance in accordance with [MIL-STD-810](#) and receive a rating of 0 or 1.
- d. Materials are treated to prevent fungus growth in a manner that does not adversely affect the unit's performance or service life and are protected from environments that would leach out the protective agent.

6.4 Processes

6.4.1 Welding

Welding of GS shall be in accordance with [NASA-SPEC-5004](#), with the exception of pressure vessels (see [5.2.13](#)).

The selection of parent materials and weld methods for a design should consider the weldments, including adjacent heat-affected zones, because they affect the operational capability of the parts concerned.

6.4.2 Brazing

Brazing should be conducted in accordance with [AWS C3.3](#).

- a. Brazing of aluminum alloys shall meet the requirements of [AWS C3.7M/C3.7](#).
- b. Torch, induction, and furnace brazing shall meet the requirements of [AWS C3.4M/C3.4](#), [AWS C3.5M/C3.5](#), and [AWS C3.6M/C3.6](#).
- c. Subsequent fusion welding operations in the vicinity of brazed joints or other operations involving high temperatures that might affect the brazed joint shall be prohibited unless it can be demonstrated that the fixturing, processes, methods, and/or procedures employed will preclude degradation of the brazed joint.
- d. Brazed joints shall be designed for shear loading and, for structural parts, not be relied upon for strength in axial loading.
- e. The shear strength of brazed joints shall be evaluated in accordance with [AWS C3.2M/C3.2](#).
- f. For furnace brazing of complex configurations, such as heat exchangers and cold plates, destructive testing shall be conducted on preproduction brazed joints to verify that the brazed layer that extends beyond the fillet area is continuous and forms a uniform phase.

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- g. Brazing of pressure vessels shall be in accordance with [ASME BPVC VIII](#).

6.4.3 Soldering

- a. Soldering of electrical connections for surface mount technology shall be in accordance with [NASA-STD-8739.2](#).
- b. Soldering of electrical connections shall be in accordance with [NASA-STD-8739.3](#).
- c. All solderable platings and protective finishes based on tin shall contain a minimum lead content of 3% by weight.
- d. For critical GS, lot sampling shall be used to verify the presence of at least 3% lead.
- e. Soldering shall not be used for structural applications.

6.4.4 Heat Treating and Plating

- a. Heat treatment of aluminum alloy parts shall meet the requirements of [SAE AMS 2770](#), [SAE AMS 2771](#), or [SAE AMS 2772](#), as appropriate.
- b. Steel parts shall be heat-treated to meet the requirements of [SAE AMS-H-6875](#) or [SAE AMS 2759](#).
- c. Heat treatment of titanium and titanium alloy parts shall meet the requirements of [SAE AMS-H-81200](#).
- d. Heat treatment of nickel-based and cobalt-based alloy parts shall meet the requirements of [SAE AMS 2774](#).
- e. Electrodeposited nickel plating shall be applied according to the requirements of [SAE AMS 2403](#) or [SAE AMS 2423](#).
- f. Electroless nickel plate shall be applied in accordance with [SAE AMS 2404](#).
- g. The nickel-aluminum interface in nickel-plated aluminum shall be protected from exposure to corrosive environments.

Nickel and aluminum form a strong galvanic cell at the nickel-aluminum interface, and exposure of the aluminum alloy to a corrosive environment can rapidly debond the nickel plate.

- h. Galvanized (zinc) coatings shall be applied in accordance with [ASTM A123](#), [ASTM A153](#), and [ASTM A653](#).
- i. All repairs to damaged galvanized coatings shall be in accordance with [ASTM A780](#).
- j. Chromium plating shall be in accordance with [SAE AMS 2460](#).

- k. Cadmium plating shall be in accordance with [SAE AMS-QQ-P-416](#).

6.4.5 Forging

For critical GS, after the forging technique (including degree of working) is established, the first production forging shall be sectioned to show the grain-flow patterns and to verify mechanical properties.

- a. The procedure shall be repeated after any change in the forging technique.

The information gained from this effort should be used to redesign the forging technique as necessary.

- b. The resulting data shall be retained and made available for review by the procuring activity.

Forging techniques should produce an internal grain-flow pattern parallel to the principal stresses, because mechanical properties are optimum in the direction of material flow during forging. The forging pattern should be free from reentrant and sharply folded flow lines.

6.4.6 Casting

Castings shall meet the requirements in [SAE AMS 2175](#).

6.4.7 Adhesive Bonding

- a. Structural adhesive bonding shall be in accordance with [MSFC-SPEC-445](#), with the exception of retesting.

Retesting of adhesives used for production parts is not required if they are within the manufacturer's recommended shelf life.

- b. Structural adhesive bonding processes shall be controlled by a documented process to prevent contamination.

The sensitivity of structural adhesive bonds to contamination is of particular concern. In the absence of relevant performance data, bond sensitivity studies should be conducted to verify that the required adhesive properties are maintained after exposure to the expected materials at the expected concentrations, including ozone, ambient humidity, cleaning fluids, and lubricants. Adequate cleanliness inspections should be conducted during the fabrication part of the bonding process.

- c. Bonded primary structural joints shall demonstrate cohesive failure modes in shear at ambient temperature.
- d. Adhesives shall not have a corrosive effect on other materials when exposed to conditions normally encountered in service.

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6.4.8 Fluid System Cleanliness

- a. Surface cleanliness levels, test methods, cleaning and packaging requirements, and protection and inspection procedures for piping, tubing, fittings, and other fluid system components shall be in accordance with [KSC-C-123](#) or as specified in the program contamination control requirements or fluid procurement and use control specification.
- b. The cleanliness level and test method shall be specified in the design documentation.
- c. For GS interfaces with precision-cleaned fluid systems for flight, the supply interface/final filters shall be located as close to the flight hardware interface as possible.
- d. Interface filters shall be used on outlet lines if it is determined that any operations, such as the servicing or deservicing of fluids, could permit flow in a reverse direction.
- e. Interfacing fluid system GS shall be cleaned to meet or exceed the cleanliness level of the flight hardware.

6.4.9 Riveting

Riveting should be in accordance with [MSFC-STD-156](#), Riveting, Fabrication and Inspection, Standard for. Though [MSFC-STD-156](#) is not current, it does contain the best practices for riveting.

6.4.10 Crimping

Crimping shall be in accordance with [NASA-STD-8739.4](#).

6.4.11 Potting and Molding

Potting and molding of electrical connectors shall be in accordance with [KSC-STD-132](#).

6.4.12 Electrical-Cable Design and Fabrication

Electrical cables shall be designed and fabricated in accordance with [KSC-GP-864](#) as determined by application, criticality, and operational environment.

6.4.13 Conformal Coating

Where electronic circuitry can be damaged or degraded by moisture, dust, chemicals, temperature extremes, mechanical stress, or vibration, conformal coating shall be required in accordance with [NASA-STD-8739.1](#).

When established, requirements for conformal coating should be stated in the design documentation.

6.4.14 Corrosion Control

Protective coating of hardware should be appropriate to the condition, use, and environment to which the hardware will be exposed during its life cycle. The coating should minimize corrosion and its color should indicate its use (see 4.2.2.3). Guidelines for corrosion control for facilities, systems, and equipment are given in TM-584.

- a. Protective coating of hardware shall be in accordance with [NASA-STD-5008](#).
- b. Corrosion control of galvanic couples shall be in accordance with [MIL-STD-889](#).
- c. All contacts between graphite-based composites and metallic materials shall be treated as dissimilar metal couples and sealed in accordance with [NASA-STD-5008](#).
- d. For critical GS, faying surfaces of metal alloys and electrical bonding connections, except for nickel-plated surfaces, shall be sealed in accordance with [NASA-STD-5008](#), except for nickel-plated surfaces.

6.4.14.1 Stress Corrosion Cracking (SCC)

In critical applications, metals exposed to the KSC environment shall be selected from alloys that are highly resistant to SCC as specified in [MSFC-STD-3029](#).

In noncritical applications, the designer should select from alloys that are highly resistant to SCC as specified in [MSFC-STD-3029](#). When COTS items are selected, the designer should attempt to identify the materials used and determine if an SCC failure would be critical. Materials protected from the environment, such as lubricated roller bearings and gears within a sealed gearbox, are exempt from the above requirement. The M&P authority's signature or electronic approval of the design drawings constitutes approval to use the materials of the design in the intended application.

6.4.15 Nondestructive Evaluation

- a. NDE for all components except fracture-critical metallic GS components shall be performed in accordance with [MIL-HDBK-6870](#).
- b. NDE for all fracture-critical metallic GS components shall be performed in accordance with [NASA-STD-5009](#).
- c. All NDE of welds shall be performed in accordance with the applicable welding specification.
- d. NDE of non-fracture-critical base materials shall be in accordance with [ASTM E1417](#), [ASTM E1444](#), [ASTM E1742](#), [ASTM E2375](#), [SAE AMS 2647](#), [SAE AMS-STD-2154](#), [SAE ARP 4402](#), and [SAE AS 4787](#)

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Typical NDE methods include penetrant, magnetic particle, radiographic, ultrasonic, and eddy current testing. NDE inspection is not limited to these methods and may include additional methods such as leak testing, and advanced methods such as shearography and thermography, as required.

- f. Qualification and certification of personnel involved in NDE shall comply with [AIA/NAS 410](#).

6.4.16 Hydrogen Embrittlement

The embrittlement of a metal or alloy by hydrogen involves the diffusion of atomic hydrogen into a component subject to a manufacturing process or environmental event. The entrained hydrogen reduces the ductility of the metal and load-bearing capacity, resulting in brittle fracture at stresses below the yield stress of susceptible materials.

Hydrogen can be introduced by processes such as electroplating, phosphating, pickling, passivation, and welding. Hydrogen can also be introduced as a by-product of a corrosion reaction in which some of the hydrogen enters the metal in atomic form.

- a. Approval by the M&P organization, in the form of an MUA, or documented approval of the design drawings is required for all metallic materials used in hydrogen systems in critical GS.

Test data may have to be generated in a simulated environment to support the approval rationale. Guidelines for designing safe hydrogen systems are contained in [ANSI/AIAA/G-095](#).

- b. After electroplating, phosphating, pickling, passivation, or other processes that can introduce hydrogen, materials that are susceptible to hydrogen embrittlement shall undergo a final baking heat treatment in accordance with [SAE AMS 2759/9](#).
- c. Corrosion protection shall be in accordance with [NASA-STD-5008](#) to minimize the risk of hydrogen embrittlement due to corrosion.
- d. Welding shall be in accordance with [NASA-SPEC-5004](#) to minimize generation of hydrogen during welding.

6.4.17 Contamination Control

- a. When required by the program, a contamination control plan shall be generated in accordance with [ASTM E1548](#).
- b. Engineering documentation shall specify internal and external surface cleanliness levels for all GS.

APPENDIX A. MATERIAL USAGE AGREEMENT

KSC Form 21-475, Material Usage Agreement, can be accessed on the Internet at <http://kscforms.ksc.nasa.gov/KSCForms2/ref/KSC21-475.itr>.

Figure 2 is an example of the first page of the agreement, which will open up in Form Filler when you go to the Internet address noted above.

Material Usage Agreement			
1. MUA No.		2. System/Subsystem	
3. Prepared By: Name		Organization	Date
4. System Dwg. No. and Title			
5. Component Specification		Vendor	P/N
6. Material and Specification		7. Material Manufacturer	
8. Use Environment			
<input type="checkbox"/> Controlled Interior <input type="checkbox"/> Indoor <input type="checkbox"/> Outdoor <input type="checkbox"/> Outdoor Launch			
Pressure _____		Temperature _____	Media _____
9. Application and Deviation			
10. Acceptance Rationale			
11. System Engineer/Lead Designer			
Name _____		Organization _____	Date _____
		Signature _____	Date _____
12. NASA Safety Representative <input type="checkbox"/> Approve <input type="checkbox"/> Disapprove			
Name _____		Organization _____	Date _____
		Signature _____	Date _____
OR			
Contractor Safety Representative <input type="checkbox"/> Approve <input type="checkbox"/> Disapprove			
Name _____		Organization _____	Date _____
		Signature _____	Date _____
13. Center Materials Representative <input type="checkbox"/> Approve <input type="checkbox"/> Disapprove			
Name _____		Organization _____	Date _____
		Signature _____	Date _____

KSC FORM 21-475 NS (REV. 12/08) PREVIOUS EDITIONS ARE OBSOLETE

Figure 2. Sample Material Usage Agreement

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APPENDIX B. DELIVERABLES**B.1 GS Documentation Deliverables**

B.1.1 The GS provider is responsible for submitting documentation to verify that the hardware/software has been developed in accordance with this standard.

B.1.2 The GS provider is responsible for providing all the necessary documentation to the using organization when the GS is delivered for use, regardless of who “owns” the GS at the time of delivery. Examples of this documentation include, but are not limited to, the following:

- Certification Approval Request (indicates how the GS was certified as complying with this standard)
- Material Inspection and Receiving Report
- Validation and verification compliance records
- Drawings with parts list or bills of material
- Maintenance manuals/procedures
- Hazard Analyses or Ground Safety Data Pack
- Operating manuals/procedures
- Design Verification Matrix (indicates which GS requirements were met and how)
- Software Version Description Document
- Facility and flight vehicle interface requirements
- Firmware Version Description Document
- Material certifications and lot traceability
- Failure Modes, Effects, and Criticality Analysis
- Critical Items List

Intent/Rationale: The using organization requires documentation for safely operating, maintaining, and servicing the GS. To reduce risk to the mission, as well as to ground personnel and flight crews, a Failure Modes and Effects Analysis should be completed and submitted in accordance with the criticality assigned to the GS by the responsible program or project. [KDP-P-2713](#) lists typical documentation requirements.

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APPENDIX C. REFERENCE DOCUMENTS

AIAA R-100	Recommended Practice for Parts Management
ANSI/AIAA/G-095	Guide to Safety of Hydrogen and Hydrogen Systems
ASME B31.1	Power Piping
ASTM D5363	Standard Specification for Anaerobic Single-Component Adhesives (AN)
ASTM G63	Standard Guide for Evaluating Nonmetallic Materials for Oxygen Service
ASTM G88	Standard Guide for Designing Systems for Oxygen Service
ASTM G94	Standard Guide for Evaluating Metals for Oxygen Service
AWS C3.3	Recommended Practices for Design, Manufacture, and Examination of Critical Brazed Components
ECA EIA/ECA 310-E	Cabinets, Racks, Panels, and Associated Equipment
IEC 62305	Protection Against Lightning, Part 1: General Principles, Part 2: Risk Management, Part 3: Physical Damage and Life Hazards, and Part 4: Electrical and Electronic Systems with Structures
IEEE 446	Recommended Practice for Emergency and Standby Power Systems for Industrial and Commercial Applications
IEEE 484	Recommended Practice for Installation Design and Installation of Vented Lead-Acid Batteries for Stationary Applications
IEEE 1106	Recommended Practice for Installation, Maintenance, Testing, and Replacement of Vented Nickel-Cadmium Batteries for Stationary Applications
IEEE 1187	Recommended Practice for Installation Design and Installation of Valve-Regulated Lead-Acid Storage Batteries for Stationary Applications
JSC 29353	Flammability Configuration Analysis for Spacecraft Applications
KDP-P-2713	Technical Review Process

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KTI-5210	Material Selection List for All Oxygen and Air Services
KTI-5212	Material Selection List for Plastic Films, Foams, and Adhesive Tapes
MIL-HDBK-454	General Guidelines for Electronic Equipment
MSFC-STD-156	Riveting, Fabrication and Inspection, Standard for
NASA/CR-2005-213424	Lubrication for Space Applications
NASA RP-1228	Fastener Design Manual
NASA/TM-2007-213740	Guide for Oxygen Compatibility Assessments on Oxygen Components and Systems
NASA-TM-86556	Lubrication Handbook for the Space Industry, Part A: Solid Lubricants, Part B: Liquid Lubricants
NASA/TP-2003-212242	EEE-INST-002: Instructions for EEE Parts Selection, Screening, Qualification, and Derating
NEMA C50.41	American National Standard for Polyphase Induction Motors for Power Generating Stations
NEMA ICS 2	Industrial Control and Systems Controllers, Contactors, and Overload Relays Rated 600 Volts
NEMA ICS 61800-2	Adjustable Speed Electrical Power Drive Systems, Part 2: General Requirements – Rating Specifications for Low Voltage Adjustable Frequency A.C. Power Drive Systems
NEMA MG 1	Motors and Generators
NPD 8730.2	NASA Parts Policy
SSP 50004	Ground Support Equipment Design Requirements: International Space Station
SW-E-0002	Ground Support Equipment General Design Requirements: Space Shuttle
TM-584	Corrosion Control and Treatment Manual

STANDARDIZATION DOCUMENT IMPROVEMENT PROPOSAL

INSTRUCTIONS

1. The preparing activity must complete blocks 1, 2, 3, and 8. In block 1, both the document number and revision letter should be given.
2. The submitter of this form must complete blocks 4, 5, 6, and 7.
3. The preparing activity must provide a reply within 30 days from receipt of the form.

NOTE: This form may not be used to request copies of documents, nor to request waivers or clarification of requirements on current contracts. Comments submitted on this form do not constitute or imply authorization to waive any portion of the referenced document or to amend contractual requirements.

I RECOMMEND A CHANGE:	1. DOCUMENT NUMBER KSC-DE-512-SM Revision L	2. DOCUMENT DATE February 29, 2012
3. DOCUMENT TITLE Facility Systems, Ground Support Systems, and Ground Support Equipment General Design Requirements		
4. NATURE OF CHANGE <i>(Identify paragraph number and include proposed rewrite, if possible. Attach extra sheets as needed.)</i>		
5. REASON FOR RECOMMENDATION		
6. SUBMITTER		
a. NAME <i>(Last, First, Middle Initial)</i>	b. ORGANIZATION	
c. ADDRESS <i>(Include Zip Code)</i>	d. TELEPHONE <i>(Include Area Code)</i>	7. DATE SUBMITTED
8. PREPARING ACTIVITY		
a. NAME Engineering and Technology Directorate, (NE)	b. TELEPHONE <i>(Include Area Code)</i> (321) 867-7770	
c. ADDRESS <i>(Include Zip Code)</i> John F. Kennedy Space Center Kennedy Space Center, Florida 32899		