MILITARY SPECIFICATION SHEET

STARTER, AIRCRAFT ENGINE, AIR TURBINE, MODEL A-24-1

The complete requirements for procuring the air turbine starter described herein shall consist of this document and the issue in effect of Specification MIL-S-19557C (AS).

STARTER, AIRCRAFT ENGINE, AIR TURBINE MOL A-24-1

APPLICABLE PARACRAPH OF SPECIFICATION	
3.8.1	Envelope: Figure 1
3.7	Wt, Max (Lb): 30
6.3.1	Aircraft Model: P-3
4.5.16	Engine Model: T56
3.8.2.1	Quick-Attach-Detach Mounting: Yes
3.8.2.2	Engine Accessory Drive: AND20002, Type XIIS
3.8.3	Air Inlet and Exhaust Connections: Figure 1
3.5.5 3.9.1	Rotation Viewed from Anti-Drive End: Clockwise
3.3.2	Exposure Temperature Range (OF): -100 to +275
3.3.2	Ambient Temperature Operating Range (OF): -65 to +160
3.3.2.3 4.5.1.1 4.5.6	Control Valve: Opening Rate, Max: 22 psig per sec for the first .1 sec Closing Time Max: 0.5 sec Navy Model: ATSCV-1
3.5.1	Rated Conditions: Output Drive Speed (RPM): 6100 Output Drive Torque, Min (Lb-Ft): 57 Airflow, Max (Lb per Min): 110 Air Inlet Total Pressure (In. Hg A): 100 Air Inlet Total Temperature (F): 350
3.5.1 3.5.7	Cutout Speed (RPM): 8500 <u>+</u> 250
3.5.7	Automatic Shutoff Control: Either turbine wheel speed or output drive speed may be sensed for control operation.

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3.5.8	Altitude: The starter rated speed and ou at rated inlet con not be damaged when tude in the overru	itput torque at iditions. The s in subjected to	8000-ft altitude tarter shall 75,000-ft alti-
3.5.9	Attitude: The starter the operational retion when the starth the output shaft, specified otherwise	equirements of t ter centerline, is in any posit	his specifica- taken through ion, unless
4.5.4	Initial Calibration:		
	Drive Torque (Lb-Ft)	Drive Speed (RPM)	Drive Speed Condition
	120 Min, 185 Max	0 6100	*Stall Rated

*If stall torque is below 158 lb-ft, starter must produce a minimum of 45 lb-ft at 8150 rpm.

8250 Min

Cutout

	Airflow, Max (Lb per Min)	Drive Speed (KPM)	Drive Speed Condition
	110	0	Stall
4.5.5	No-Load Operation: Time (Sec): 60 Minimum Drive Spe	ed (RPM): 9500	
3,3,5 4.5.6		per of cycles: 12 ge permitted. Eac consecutive phase	ch cycle shall

Test Stand Rotor Polar		e A - ration	At Foll	Phase B owing Cond		Phase <u>Acceler</u>	_
Moment of Inertia	Drive Speed	Time	Drive Speed	Drive Torque	<u>Time</u>	Drive <u>Speed</u>	<u>Time</u>
Lb-Ft ²	RPM <u>+50</u>	Max Sec	RPM <u>±50</u>	Lb-Ft (Min)	Sec	Min RPM	Sec
72 * 72	2475 2850	8 8	2475 2850	115 	10 10	6500 6500	12 12

^{*}Last 600 cycles only; run at 650°F air inlet temperature.

APPLICABLE PARAGRAPH OF SPECIFICATION

The above values do not allow for flywheel windage losses, test stand drag, or inertia or the starter. In order to determine compliance with this specification, the test equipment must be calibrated and specific test results corrected accordingly. If the acceleration specified for Phase C is not developed because of test standard torque, the test standard inertia load may be reduced or external power may be applied to assist acceleration to cutout speed. Time from end of Phase C to starter cutout shall not exceed 15 seconds.

4.5.8	Rated Torque Calibration:
	Minimum Output Drive Speed (RPM): 6100
	Airflow, Max (Lb per Min): 110
2.2 :	0

3.3.5 Overrunning: The starter output drive shall be driven for 10 cycles in a clockwise direction (viewed from the anti-output shaft end) in accordance with the following 100-hour cycle:

Position of Starter Anti-Drive End Will Respect to Starter Horizontal Centerline +2°	Duration of Running Hours	Output Shart Speed RPM ±250
-15°	6	14,230
_ 5 ^o	42	14,230
() (horizontal)	50	14,230
+100	2	10,510

4.5.10 Extreme Temperature Operation:

	Inlet A	ir Conditions		
	Total Press. In. ag A ±0.5		Ambiert Temp (OF)	
	140	250 44.5	-65 +160	
4.5.11	Consecutive Cyclical	Operation: M	unher of cycles:	5
4.5.12	Sustained Totoring:	Time (!linutes.); 2	
4.5.13	Vibration: MIL-STD-8 Curve F (Parts 1,		4. Procedure I,	
4.5.15.1	Slip Cluter Torque, M	ax (Lt-Ft): A	(no	
4.5.15.2	Slear Section Strengt	h, Max (Lb-Ft)); 400	

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4.5.18 Containment Tests: Applicable

4.5.18.1 HUB Containment Tests

Inlet Air Pressure (PSIA): 49

Inlet Air Temperature (°F): 345

4.5.18.2 Failure at No-Load:

Inlet Air Pressure (PSIA): 110

Inlet Air Temperature: (°F): 680

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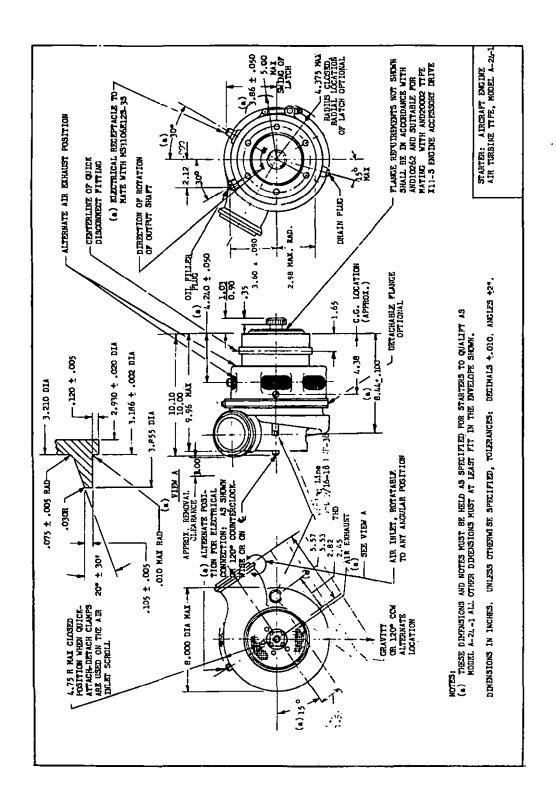


Figure 1

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SPECIFICATION ANALYSIS SH	Form Approved Budget Bureau No. 22-R255		
INSTRUCTIONS: This sheet is to be filled out by personnel, either Government or contractor, involved in the use of the specification in procurement of products for ultimate use by the Department of Defense. This sheet is provided for obtaining information on the use of this specification which will insure that suitable products can be procured with a minimum amount of delay and at the least cost. Comments and the return of this form will be appreciated. Fold on lines on reverse side, staple in corner, and send to preparing activity. Comments and suggestions submitted on this form do not constitute or imply authorization to waive any portion of the referenced document(s) or serve to amend contractual requirements.			
MIL-S-19557/7(AS) STARTER: AIRCRAF	T ENGINE, AIR TURE	BINE	
ORG ANIZATION			
CITY AND STATE	CONTRACT NUMBER		
	CONTRACT		
1. HAS ANY PART OF THE SPECIFICATION CREATED PI MENT USE? A. GIVE PARAGRAPH NUMBER AND WORDING.	ROBLEMS OR REQUIRED INT	ERPRETATION IN PROCURE-	
A. GIVE PARAGRAPH NUMBER AND WORDING.			
B. RECOMMENDATIONS FOR CORRECTING THE DEF	CIENCIES		
2. COMMENTS ON ANY SPECIFICATION REQUIREMENT CONSIDERED TOO RIGID			
3. IS THE SPECIFICATION RESTRICTIVE?			
YES NO (II "yes", in what way?)			
4. REMARKS (Attach any pertinent data which may be of use in improving this specification. If there are additional papers, attach to form and place both in an envelope addressed to preparing activity)			
		~	
SUBMITTED BY (Printed or typed name and activity - Option	nal)	DATE	